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# STRUCTURAL AREA INSPECTION FREQUENCY EVALUATION (SAIFE)

Volume IV. Software Documentation and User's Manual **Book 1. Initial Program** 

> Carter J. Dinkeloo Martin S. Moran





**APRIL 1978 FINAL REPORT** 

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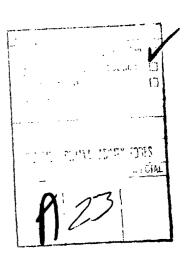
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### PREFACE

Technology Incorporated prepared this fourth volume of a five-volume report to document the simulation logic for the Structural Area Inspection Frequency Evaluation (SAIFE) in accordance with Article II, paragraph B of Contract DOT-FA74WA-3493. (Volume IV along with Volume V completes the requirements of Phase III of the contract.) The effort is sponsored by the Aircraft Safety and Noise Abatement Division, Systems Research and Development Service of the Federal Aviation Administration.

The principal Technology Incorporated personnel engaged on this program were Mr. Carter J. Dinkeloo, project engineer, who served as principal investigator; Mr. Martin S. Moran, research engineer, who developed the model for the SAIFE computer program; and Mr. Ronald I. Rockafellow, program manager.

The contract monitors for the FAA were Messrs, Herbert Spicer and Charles Troha of the Aircraft Safety and Noise Abatement Division. The technical monitor was Mr. Arnold E. Anderjaska of the Flight Standards Division.



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### I. INTRODUCTION

It is the mutual goal of the FAA, airframe manufacturers, and air carriers to constantly improve the structural integrity and inspection efficiency of civil aircraft. The good safety record of U.S. air carriers indicates that the current process of establishing and modifying structural inspection programs has been successful. However, with the increasing size and complexity of second- and third-generation transport aircraft, there is a need to quantify more precisely the present, subjective evaluation process which relies heavily on reliability analyses of the new design and on operational experience of similar aircraft.

Because of the extreme complexity of the evaluation process, a computer simulation of all critical aircraft service life aspects was judged the most rational means for quantifying the process more exactly. As a five-volume document, this report documents the resultant Structural Area Inspection Frequency Evaluation (SAIFE) simulation logic. SAIFE accounts for the following factors: (1) aircraft design analysis; (2) component and fullscale fatigue testing; (3) production, service, and corrosion defects; (4) probability of crack or corrosion detection; and (5) aircraft modification econimics. It treats these factors in a logical sequence that realistically represents the procedure currently used to establish and modify inspection intervals. Figure 1 illustrates the data sources and analytical functions that are integrated into the SAIFE logic. SAIFE is designed to provide a repeatable method for evaluating proposed inspection programs. However, it is not intended to supplant the Maintenance Review Board or the air carrier use of the Standard Operations Specification - Aircraft Maintenance.

As Volume IV, this user's manual for the SAIFE program contains a system description, a program description, operating procedures, a sample input and output, and a source listing of the program. The detailed description of the program events and routines is presented in Appendix A, and the program source listing is contained in Appendix B which because of its voluminous computer-generated data is on the included microfiche.

The original computer model developed during the initial contract has been modified by the Engineering and Manufacturing Branch, Flight Standards National Field Office for the parametric study and is documented in Book 2 of this volume.

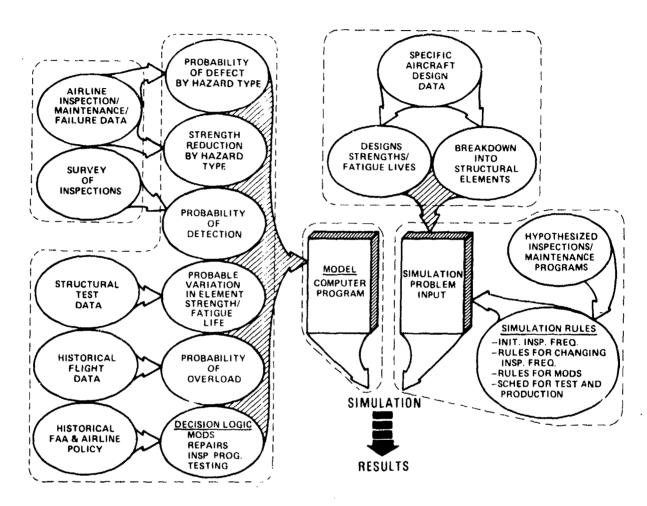


Figure 1. Approach to SAIFE Simulation

### II. SYSTEM DESCRIPTION

The eight blocks in Figure 2 represent the major aspects of the SAIFE simulation logic. Block 1 accepts input data for the aircraft fleet and for each structural element in the aircraft. After determining whether element modifications are required because of the fatigue test results in Block 2, Block 1 assigns a fatigue life to each element in each aircraft. Block 3 determines whether production, service, or corrosion defects will occur; if it is determined that such defects will occur, Block 3 predicts the times when they will occur. After comparing the flight loads with the strength of each element, Block 4 predicts the time to failure for each element. Block 5 conducts the periodic inspections of each element. If defects are detected, Block 6 repairs the element and assigns it a new fatigue life. However, if an existing defect is allowed to grow until element failure, Block 5 deletes the aircraft from the fleet. Depending on the magnitude of the detected defects, special inspections and increased inspection frequencies may be called for in Block 7 and modifications may be instituted in Block 8. When all the aircraft have been deleted from the fleet or retired from service, the simulation is complete.

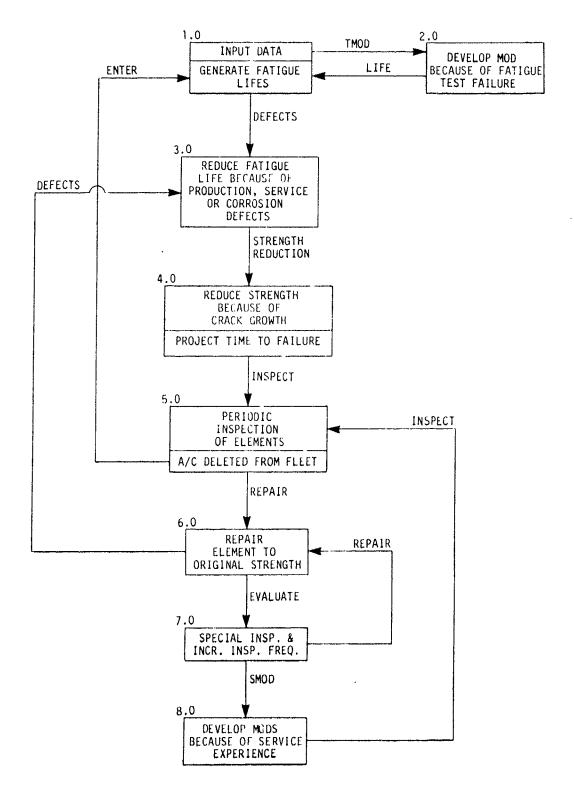


Figure 2. Major Aspects of SAIFE Logic

### III. GENERAL PROGRAM DESCRIPTION

The SAIFE program is a large, complex math model designed to simulate the structural performance of aircraft in a fleet of aircraft and the effectiveness of the inspection program for the aircraft fleet. The aircraft model included in the math model is divided into structurally significant elements and the inspection program for each element is defined. Structural defects are classified as follows: fatigue and corrosion which are wear-out and aging phenomena; production or design defects; and operational or maintenance damage. These defects and the inspection program are treated as probabilistic phenomena interacting over time. If the simulation is to proceed properly, the passage of simulated time must be controlled. This control can be accomplished by a user-designed algorithm or it can be done automatically with a "simulation clock" in one of the special purpose simulation languages. SIMSCRIPT II.5  $^{\rm T}$  (trademark, Consolidated Analysis Centers Inc., Los Angeles, California), a computer language designed for discrete-event simulation applications, was chosen for the SAIFE simulation. SIMSCRIPT II.5 is a large language designed to facilitate the simulation of large, complex systems, and to reduce the total time spent in designing, programming, and testing simulation models.

Because of the extensive detail, the program events and routines are described in Appendix A.

## IV. OPERATING PROCEDURES

# 1. Computer Requirements

The SAIFE demonstration output presented in Volume V of this report was run on an IBM Model 300-05 computer using the SIMSCRIPT II.5 compiler, release 8F. The execution time per element ranged between 12 and 20 cpu seconds, and the compilation time was 205 cpu seconds. None of the demonstration runs required more than 325K bytes of core storage.

The run time and the core storage requirements depend on the values of the input variables. The variable that probably has the greatest effect on these requirements is the actual average fatigue life. The shorter the life, the greater the requirement: A shorter actual average fatigue life increases crack initiations across the floot. These additional crack initiations, in turn, increase the inspection process, the repair activities, and the modification and interval change decisions. All these additional events require more execution time and storage space for event notices. The actual average fatigue life is shortened when the parameters involved in the distribution of the ratio of the actual to the predicted average fatigue life are so changed that the ratio is decreased.

Other variables which affect the run time and core storage requirements are as follows: with the actual average fatigue life established for all aircraft in the fleet, the requirements are increased if the parameters involved in the Weibull distribution of actual fatigue lives for individual aircraft are so changed that some lives are shortened and more crack initiations are introduced. In addition, regardless of the fatigue lives, the run time and storage requirements are increased if the defect occurrence rates are increased or the initial inspection intervals are decreased.

Since the seeds of the random number streams are initialized to the same values at the start of each execution, the user can reproduce the output, if desired, and more easily identify effects because of changes in the input parameters. Table 1 lists the ten seeds used by the IBM 360-65 to generate the demonstration output presented in Volume V.

### 2. Input

The program input consists of three parts. The first part contains input variables which pertain to the aircraft type under consideration. These variables are input only once per simulation run and are constant from element to element. The second part is optional and is explained in Section IV, 2.2. The third part contains input variables whose values are unique to each element. These variables must be input in their entirety for each element being simulated.

### TABLE 1. INITIAL RANDOM ROOTS

SEED.V(1) = 2116429302

SEED.V(2) = 683743814

SEED.V(3) = 964393174

SEED.V(4) = 1217426631

SEED.V(5) = 618433579

SEED.V(6) = 1157240309

SEED.V(7) = 15726055

SEED.V(8) = 48108509

SEED.V(9) = 1797920909

SEED.V(10) = 477424540

### 2.1 Aircraft Data

The input variables which pertain to the aircraft type are listed and described below in the order in which they are read in by SAIFE.

MODEL(\*) - This one-dimensional alpha array of size two identifies the aircraft type under consideration. The total length of this identification cannot exceed eight characters.

SIZE.OF.FLEET - This integer variable is the number of aircraft in the fleet being simulated. The output format requires that this variable does not exceed 99999.

USAGE.LIFE - This real variable is the service life in flight hours of the aircraft being simulated. All aircraft in the fleet must have the same service life. The output format requires that this variable does not exceed 9999999.

BEGIN.PRODUCTION - This real variable is the time in flight hours relative to the start of the simulation when the first aircraft enters service. This variable in conjunction with the input variable START.TEST enables the user to start the fatigue test of the element before, after, or at the same time the first aircraft enters service.

PRODUCTION.TIME - This real variable defines the initial aircraft production rate. It is the time in flight hours between aircraft entering service.

2.PRODUCTION.TIME - This real variable defines the second aircraft production rate. It is the time in flight hours between aircraft entering service.

- PRCHG This real variable is the simulation time when the second aircraft production rate takes effect. Note that this time is measured from the time that the first aircraft enters service and not from the start of the simulation.
- START. TEST This real variable is the time in flight hours relative to the start of the simulation when the fatigue test of an element is begun. If no fatigue test is to be conducted, this variable is set to the machine upper limit.
- TEST.ACCEL.FACT This real variable is the fatigue test acceleration factor, that is, the quotient of the equivalent flight hours divided by the fatigue test hours.
- C.GROWTH.RATE This real variable is the corrosion area growth rate in sq. inches per hour for the aircraft being considered. The growth rate for each element in the aircraft is modified by its associated CRR (corrosion resistance rating).
- C7 If a modification is developed because of a fatigue test failure, this real variable is the percentage (expressed as a decimal fraction) of the test life when the inspection frequency is increased.
- C28 This real variable is the percentage (expressed as a decimal fraction) reduction in the remaining fatigue life of an element when corrosion occurs in a stress concentration.
- C29 This real variable is the percentage (expressed as a decimal fraction) reduction in the remaining fatigue life of an element when corrosion occurs outside a stress concentration.
- MU.R This real variable is the mean of the log-normal distribution of the ratio of the actual average fatigue life to the predicted average fatigue life.
- SIG.R This real variable is the standard deviation of the log-normal distribution of the ratio of the actual average fatigue life to the predicted average fatigue life.
- $\underline{A}$  This real variable is the result of fitting an exponential curve to flight load exceedance data. Aexp[BSa] is the number of flight loads per hour which exceed the load level Sa.
- $\frac{B}{t}$  This real variable is the result of fitting an exponential curve to flight load exceedance data. Aexp[BSa] is the number of flight loads per hour which exceed the load level Sa.

<u>1ABCD(\*)</u> - This one-dimensional real array of size four contains the initial lengths in flight hours of the inspection intervals of the four levels of scheduled inspections. 1ABCD(1) corresponds to the A-level interval; 1ABCD(2) corresponds to the B-level interval; 1ABCD(3) corresponds to the C-level interval; and 1ABCD(4) corresponds to the D-level interval.

CABCD(\*) - This one-dimensional real array of size four contains the inspection cost at each level of inspection. CABCD(1) corresponds to the A-level cost; CABCD(2) corresponds to the B-level cost; CABCD(3) corresponds to the C-level cost; and CABCD(4) corresponds to the D-level cost.

SAMPLING - This real variable is the percentage of the fleet to be sampled during a D-level inspection.

LONG.LIST - This alpha variable is read in as "YES" when the long list output option is desired; otherwise, it is read in as "NO."

PCCL - This real variable is the percentage (read in as a decimal fraction) of the element critical crack length at which a crack, which initiated internally, becomes external.

### 2.2 Long List Data

Occasionally in the standard output, elements will appear with unusually long fatigue cracks or early element failures. It is desirable to have a more complete service history of aircraft with these early element failures than that offered by the standard output. This service history is available through what is called the long list option. This output option is accessed by reading in alpha characters "YES" for the aircraft input variable LONG.LIST. After this input, the element description and identification numbers of the aircraft to be tracked are read in. The input variables for the long list option are listed and described below in the order in which they are read in by SAIFE.

NOE - This integer variable is the number of elements to be processed under the long list option.

ELID(\*,\*) - This two-dimensional alpha array of size four by NOE identifies each element to be processed. This identification must appear in the first sixteen columns of the data card and must be identical to the description read into the variable ELEMENT(\*) described in Section IV, 2.3.

NOAC(\*) - This one-dimensional integer array of size NOE is the number of aircraft to be tracked for each corresponding element.

TLID(\*,\*) - This two-dimensional integer array of size NOE by NOAC(\*) contains the identification numbers of the aircraft to be tracked for a particular element.

The above data are read in immediately following the aircraft input data only when the long list option is desired. Normally, the long list option will be used only after a standard run has indicated a problem area. Since the SAIFE program depends on many sequences of random numbers, all elements in the long list run must be in the same sequence as those in the first run. When the long list option is in effect, the standard output is suppressed.

### 2.3 Element Data

The input variables which are unique to each element and must be read in for each element are listed and described below in the order in which they are read in by SAIFE.

ELEMENT(\*) - This one-dimensional alpha array of size four identifies the element being simulated. The total length of this identification cannot exceed sixteen characters.

PREDICTED.LIFE - This real variable is the average element fatigue life in flight hours predicted by analysis. If the actual average fatigue life is known, this variable can be entered as zero. The output format requires that this variable does not exceed 999999.

ACTUAL.AVG.FAT.LIFE - This real variable is the actual average element fatigue life in flight hours determined by fatigue test. If this value is not known before running the simulation, input zero and SAIFE will determine it statistically.

M1.MEAN - This real variable is the average slow crack propagation rate for the element being simulated.

M2.MEAN - This real variable is the average fast crack propagation rate for the element being simulated.

LGHT.TO.FAILURE - This real variable contains the element crack length which corresponds to zero residual strength or level-flight structural failure.

CRIT.CRK.LGT - This real variable is the element critical crack length, that is, the crack length when the crack propagation rate changes from slow to fast.

FSAF.LGT - This real variable is the element fail-safe crack length.

BIRTH. DEFECT. PROBABILITY - This real variable is the probability (expressed as a decimal fraction) that the element has a production defect when the aircraft enters service.

<u>CRR</u> - This integer variable is the element Corrosion Resistance Rating.

- SDM. OCCURRENCE. RATE This real variable is the service damage occurrence rate in occurrences per element per aircraft per flight hour.
- LEAD.TIME This real variable is the time in flight hours between the time when the decision is made to develop a modification and the time when the modification is ready for implementation.
- T.FREQ.CHG This real variable is the factor (expressed as a decimal fraction) by which inspection intervals are decreased because of a fatigue test failure.
- S.FREQ.CHG This real variable is the factor (expressed as a decimal fraction) by which inspection intervals are decreased because of unfavorable service experience.
- FREQ.DECREASE This real variable is the factor (expressed as a decimal fraction) by which inspection intervals are increased because of favorable service experience.
- 1.PROB Given that there is fatigue crack initiation, this real variable is the probability that the crack initiates internally. This variable applies to all three allowable fatigue cracks.
- <u>C.PROB</u> Given that there is corrosion initiation, this real variable is the probability that the corrosion initiates internally.
- INT.LVL.INSP This alpha variable is the letter designation of the lowest internal inspection level.
- EXT.LVL.INSP This alpha variable is the letter designation of the lowest external inspection level.
- A.REPAIR.COST This real variable is the repair cost of a defect detected during an A-level inspection.
- B.REPAIR.COST This real variable is the repair cost of a defect detected during a B-level inspection.
- C.REPAIR.COST This real variable is the repair cost of a defect detected during a C-level inspection.
- D.REPAIR.COST This real variable is the repair cost of a defect detected during a D-level inspection.
- MOD. TESTED This alpha variable indicates whether or not a structural modification is to be fatigue tested. The two acceptable input values are YES or NO.
- 1ST.TOOLING This real variable is the tooling cost in the development of the first structural modification.

in addition the wife had a survey of the second of the

AD. TOOLING - This real variable is the tooling cost in the development of any additional structural modifications.

<u>1ST.MD.COST</u> - This real variable is the installation cost of the first structural modification.

AD. MD. COST - This real variable is the installation cost of any additional structural modifications.

S.REPAIR.COST - This real variable is the repair cost of a defect detected during a special inspection.

LOCATED. IN. STRESS. CON - This real variable is the probability (expressed as a decimal fraction) that there is corrosion in a stress concentration.

- 1.CDM.OCCURRENCE.RATE This real variable is the initial corrosion occurrence rate in occurrences per element per aircraft per flight hour.
- 2.CDM.OCCURRENCE.RATE This real variable is the second corrosion occurrence rate in occurrences per element per aircraft per flight hour.

CDM.RATE.CHANGE - This real variable is the aircraft service time in flight hours when the second corrosion occurrence rate takes effect.

# 2.4 Format Specifications

Most of the input data are entered into SAIFE by the free-form read statement. The program has only three formatted read statements.

The aircraft type identification, the alpha array MODEL, is entered under the format specification 2A4. This identification must be contained in the first eight columns of the first card of the Aircraft Input Data. All subsequent data in this section can appear in any columns and on as many cards as desired. All input values must be separated from one another by at least one blank column and a value cannot be continued on the next card.

The element identification alpha array ELID is entered under the format specification 4A4. This identification must be contained in the first sixteen columns of the long list element data card. Subsequent data can appear in any column and on as many cards as desired. When a second element is to be identified, its description must again appear in the first sixteen columns of the data card.

The element identification alpha array ELEMENT is entered under the format specification 4A4. This identification must be contained in the first sixteen columns of the first card of each set of Element Input Data. As in the Aircraft Input Data, all

subsequent data in this section can appear in any columns and on as many cards as desired.

Sample input data consisting of Aircraft Input Data and five sets of Element Input Data are illustrated in Figure 3. The aircraft type identification is HYBRID. The two cards immediately following the aircraft type card contain the Aircraft Input Data. The five elements shown in the sample are WSC-SWB-AFT-0000, WSC-SWB-AFT-0030, WSC-SWB-AFT-0060, WSC-SWB-AFT-0090, and WSC-SWB-AFT-0127. The Element Input Data for each element begins immediately after the element identification on the same card and terminates on the last card before the next element identification. The card following the last set of Element Input Data must contain EOD in the first four columns.

500 60000 150 50 100 5000 0 100 .002 .8 .20 .40 1.00 .354 .284325 -8.80901 25 200 1000 12000 1440 7080 13563 19691 .25 NO 1.0 WSC-SWH-AFT-0000 745200 0 8.00E-5 1.60F-3 140. 1.371 15.0 1.19E-5 2 .800 .650 .250 .466 .667 D B 33 78 144 208.5 YES 13.31E-9 1476 10000 10000 11584 11584 416 .056 9.154F-9 7.526E-P 7400 WSC-SWB-AFT-0030 690000 0 8.00E-5 1.60E-3 140. 1.371 15.0 1.19E-5 3 .Ann .650 .250 .466 .667 N H 33 7H 144 20A.5 YES 13.31E-9 1476 10000 10000 11584 11584 416 .056 9.154E-9 7.526E-8 7400 WSC-SWR-AFT-0060 662400 0 8.00E-5 1.60E-3 WSC-SWB-AFT-0060 662400 0 R.00E-5 1.60E-3 140. 1.371 15.0 1.19E-5 1 13.31E-9 1476 .800 .650 .250 .466 .667 D B 33 78 144 208.5 YES 10000 10000 11584 11584 416 .056 9.154E-9 7.526E-8 7400 WSC-SWR-AFT-0090 593400 0 8.00E-5 1.60E-3 140. 1.371 15.0 1.19E-5 2 .800 .450 .250 .466 .667 D B 33 78 144 208.5 YES 13.31E-9 1476 10000 10000 11584 11584 416 .056 9.154E+9 7.526E+8 7400 WSC-SWB-AFT-0127 607200 0 A.00E-5 1.60E-3 140. 1.371 15.0 1.19E-5 3 13.31E-9 1476 .800 .650 .250 .466 .667 D 6 33 78 144 208.5 YES 10000 10000 11584 11584 416 .056 9.154E-9 7.526E-8 7400

Figure 3. Sample Input Data to Produce Standard Output

Figure 4 illustrates the same data except that the variable LONG.LIST is now entered as "YES." The Long List Input data appears between the Aircraft Input Data and the Element Input Data. The Long List Input causes the program to track aircraft numbers 100, 200, and 300 for the element WSC-SWB-AFT-0000; aircraft numbers 86 and 497 for the element WSC-SWB-AFT-0030; aircraft number 1 for the element WSC-SWB-AFT-0060; aircraft numbers 9, 10, 11, and 12 for the element WSC-SWB-AFT-0090; and aircraft numbers 323, 456, and 472 for the element WSC-SWB-AFT-0127.

### 3. Output

Each element to be simulated by SAIFE is identified by three groups of alpha characters and one group of numeric characters. The alpha characters define the basic element type and general location on the aircraft, and the numeric characters define the specific location of the element by identifying the wing or fuselage station number. For example, an element identified as WNG-STR-CEN-396 would be a wing stringer located midway between the front and rear spars and centered at wing station 396.

500 60000 150 50 100 5000 0 100 .002 .8 .20. .40 1.00 .354 .284325 -8.80901 25 200 1000 12000 1440 7080 13563 19691 .25 YES 1.0 WSC-SWB-AFT-0000 100 200 300 WSC-SWB-AFT-0030 2 86 497 WSC-SWB-AFT-0060 1 WSC-SWH-AFT-0090 Q ш 10 11 323 456 472 WSC-SWH-AFT-0127 3 WSC-SWB-AFT-0000 745200 0 8.00E-5 1.60E-3 140. 1.371 15.0 1.19E=5 2 13.31E-9 1476 .800 .650 .250 .466 .667 D B 33 78 144 208.5 YES 10000 10000 11584 11584 416 .056 9.154F-9 7.526E-R 7400 WSC-SWB-AFT-0030 690000 0 8.00E-5 1.60E-3 140. 1.371 140. 1.371 15.0 1.196-5 3 13.31E-9 1476 .800 .650 .250 .466 .667 D B 33 78 144 208.5 YES 10000 10000 11584 11584 416 .056 9.154E-9 7.526E-8 7400 WSC-SWB-AFT-0060 662400 0 8.00E-5 1.60E-3 140. 1.371 15.0 1.19E-5 1 .800 .650 .250 .466 .667 D B 33 78 144 208.5 YES 13.31E-9 1476 10000 10000 11584 11584 416 .056 9.154E-9 7.526E-8 7400 WSC-SWR-AFT-0090 593400 0 P.00E-5 1.60E-3 140. 1.371 15.0 1.196-5 2 13.31E-9 1476 .800 .650 .650 .466 .667 D R 33 78 144 208.5 YES 10000 10000 11584 11584 416 .056 9.154E-9 7.526F-8 7400 WSC-SWB-AFT-0127 607200 0 8.00E-5 1.60E-3 140. 1.371 140. 1.371 15.0 1.19E-5 3 13.31E-9 1476 .800 .650 .850 .466 .667 D B 33 78 144 206.5 YES 10000 10000 11584 11584 416 .056 9.154F-9 7.526E-8 7400

Figure 4. Sample Input Data to Produce Long List Output

The standard program output consists of two parts. The first part consists of the simulation results for each specific element. This part is printed for each set of Element Input Data. The second part consists of a summary of the first parts for an element type. In the example discussed above, WNG-STR-CEN identifies the element type. Whenever the program encounters a set of Element Input Data in which any single character of the three groups of alpha characters differs from those in the previous set of Element Input Data, a summary is printed.

A third output is available as an option. This long list option gives a more complete service history of certain selected aircraft and is discussed in Section IV,2.2. When the long list option is in effect, the standard output is suppressed.

### 3.1 Element Data

Figure 5 illustrates a sample output for the input shown in Figure 3. The aircraft type identification is the aircraft input array MODEL. The number of aircraft in the fleet is the aircraft input variable SIZE.OF.FLEET. The aircraft service life is the aircraft input variable USAGE.LIFE. The structural element identification is the element input array ELEMENT. The predicted average fatigue life is the element input variable PREDICTED.LIFE. The actual average fatigue life is the element input variable ACTUAL.AVG.FAT.LIFE.

AIRCRAFT TYPE: HYBRID

TICTED AVFRAGE FAITGUE LIFE: 745200 HOURS  NUMBER AND TIME TO INITIATION OF AIRCRAFT DEFECTS  FIRST CHACK  CORROSION  SERVICE DAMAGE  52752 537720 546270  17720 5		STRU	STRUCTURAL ELEMENT: WSC+SW8-AFT+0000	SC-SW8-AFT-0000		!
CHACK  CHACK  CORPOSION  SERVICE DAMAGE  752  37720  37720  37720  46270  600  600  600  600  600  600  600	PREDICTED AVE	RAGE FATIGUE LIFE:	745200 MOURS	ACTUAL AVERAG		670108 MDURS
TESS CRACK CORPOSION SERVICE DANAGE  18772		NUMBER	AND TIME TO INITIA	TION OF AIRCRAFT DEFI		
TES 32752 37720 59789 59		ACAUT TOUTH	CORPOSION	SERVICE DAMAGE	PRODUCTION DEFECTS	EFECTS
CES NCES IN) IN) IN) ION INTERVAL		B		000000000000000000000000000000000000000	0	
ICES IN) IN) IN) ION INTERVA	CURRENCES	~		32752	•	
CES NCES IN) IN) IN) ION INTERVAL	N(HRS)	32752	37720	59789		
INTERVA	K(HRS) G(HRS)	46270	37720	0/245	•	
INTERVA	,	NUMBER AND LE	ENGTH OF CRACKS DETE	CTED AT EACH LEVEL O	IF INSPECTION	
Interval			13.31-0	C-LEVEL	D-LEVEL	SPECIAL
INTERVA		A-LEVEL	100000000000000000000000000000000000000			
INTERVAL			فين			
NCES IN) .IN) .IN) TION INTERVAL	CURRENCES		2.22	• 0	• (	
IN) IN) IN) IN) ION INTERVAL		•	2.22	•		•
INTERVAL	(NI)9(	•	2.22	•	•	
Interval		NUMBER AND AREA O	F COPROSION DEFECTS	DETECTED AT EACH LE	VEL OF INSPECTION	
INTERVALS(HRS)  25 200 1000 25 200 1250 25 200 1563 25 200 1563 25 200 684 25 25 200 684 25 25 200 684			•	1979	D-LEVEL	SPECIAL
INTERVALS(HRS)  25 200 1000 25 200 11000 25 200 1563 25 200 1563 25 200 1563 25 200 1563 25 200 1563 25 200 1563 25 200 1563		A-LEVEL	8 - F - V - F - F - F - F - F - F - F - F	) † 1 † 1 † 1 †		-
INTERVALS(HRS)  25 200 1000 25 200 1250 25 200 1953 25 200 684 25 200 854 25 200 854		) « • • • • • • • • • • • • • • • • • • •	C	0		1 21
INTERVALS(HRS)  25	CCURRENCES			•0	•	35.47
INTERVALS(HRS)  25 200 1000 25 20 1250 25 200 1953 25 200 1953 25 200 854 25 200 854 25 200 854	(21 *20) ZI	• 0	•0	<b>.</b>	• •	35.47
25 200 1000 25 20 200 1250 25 20 200 1250 25 20 200 1263 25 20 200 1068 25 20 200 1068	VG(SO.IN)	• •	•0	• 0	•	•
25	NSPECTION INTER	VALS (HRS)				
25 500 1250 25 500 1563 25 500 854 25 500 854 25 500 854	1		500	1000	12000	
25 200 1563 25 200 1963 25 200 684 25 200 854 25 200 1068	WITIAL		302	1250	00001	
25 200 1955 25 200 854 25 200 1068	~ .	<u> </u>	00≥	1563	00.001	
25 200 854 25 200 1068 25 200 1068	•	: X	200	1953	200	
25 200 854 25 200 1068	<b>.</b>	25	200	37 E	2020	
55	n 4	52	200	300	12817	
	, ~	52	002	2001		
NUMBER OF SPECIAL INSPECTIONS CONDUCTED: 1	NUMBER OF SPECIA	L INSPECTIONS CONDU		v 2		

RESIDUAL STRENGTH EQUALS FAIL-SAFE STRENGTH AIRCRAFT ND. FLT. HOURS Sample Standard Output Produced by Input Shown in Figure 3 STRUCTURAL FAILURES AIRCRAFT NO. FLT. HOURS 5 Figure

AIRCRAFT TYPE: HYBRID

60000 MOURS	
AIRCRAFT SERVICE LIFE:	
200	
IN FLEETS	
OF AIRCRAFT IN FLEET:	

0		HOURS
00 HCUK		424261
9		.IFE:
AIRCRAFT SERVICE LIFE: 60000 HUUNS		ACTUAL AVERAGE FATIGUE LIFE: 424261 HOURS
FT SE	_	AVERA
AIRCHA	STRUCTURAL ELEMENT: MSC-SWB-AFT-0030	ACTUAL
200	URAL ELEMENT:	OO HOURS
IUMBER OF AIRCRAFT IN FLEET: 500	STRUCT	PREDICTED AVERAGE FATIGUE LIFE: 690000 HOURS
NUA		PREDICTE

NUMBER AND TIME TO INITIATION OF AIRCRAFT DEFECTS

PRODUCTION DEFECTS  0	SPECIAL 0 0.00.00.	ECTION
SERVICE DAMAGE PRI	NUMBER AND LFNGTH OF CRACKS DETECTED AT EACH LEVEL OF INSPECTION  LEVEL B-LEVEL C-LEVFL D-LEVEL  0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	NUMBER AND AREA OF CORROSION DEFECTS DETECTED AT EACH LEVEL OF INSPECTION
CORROSION 	NGTH OF CRACKS DETECT B-LEVEL 0 0 0. 0.	CORROSION DEFECTS D
FIRST CRACK 1 51286 51286 51286	NUMBER AND LFI	NUMBER AND AREA OF
OCCURRENCES MIN(HRS) MAX(HRS) AVG(HRS)	OCCURRENCES Min(IN) Max(IN) Avg(IN)	

0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0		A-LEVEL B-LEVEL	B-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
25 200 1000 25 200 1250 25 200 1563 25 200 1953	OCCURRENCES MIN(SO.IN) MAX(SO.IN) AVG(SO.IN)		e 	• • • •	•••	000
25 200 1000 25 200 1250 25 200 1563 25 200 1963	INSPECTION INTER	PVALS(HRS)				
	INITIAL 2 3 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	\$\$ \$\$ \$\$ \$\$	200 200 200 200 200	1000 1250 1563 1953	12000 15000 18750 23438	

424261 HOURS 0 NUMBER OF STRUCTURAL MODIFICATIONS: 0 FINAL ACTUAL AVERAGE MODIFIED FATIGUE LIFE: NUMBER OF AIRCRAFT MODIFIED IN SERVICE: STRUCTURAL FAILURES AIF-RAFT NO. FLT. HOURS 

RESIDUAL STRENGTH EQUALS FAIL-SAFE STRENGTH AIRCRAFT NO. FLT. HOURS

16

Figure 5 - Continued

HESIDUAL STRENGTH EQUALS FAIL-SAFE STRENGTH AIRCRAFT NO. FLT. HOURS Figure 5 - Continued

AIRCRAFT TYPE: HYBRID

AIRCRAFT SERVICE LIFE: 60000 HOURS NUMBER OF AIRCRAFT IN FLEET: 500

STRUCTURAL ELEMENT: MSC-SWB-AFT-0060

ACTUAL AVERAGE FATIGUE LIFE: 547824 HOURS PREDICTED AVERAGE FATIGUE LIFF: 662400 HOURS

NUMBER AND TIME TO INITIATION OF AIRCRAFT DEFECTS

C TACKET COMPANY		c		:::
TO LIBRAL DING KARONON	H OF CRACKS DETEC	NUMBER AND LENGTH OF CRACKS DETECTED AT EACH LEVEL OF INSPECTION	INSPECTION	
	A-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
OCCURRENCES 0	; ; ; ;	; C	0	0
MIN(IN) 0. 1.	1.12	• 0	• •	o d
• 0	1.12	• 0		

		A-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
OCCURRENCES		*	O.	***	
MIN(SO.IN)	• 0	6.61	• 0	17,63	
MAX(SQ.IN)	•0	. 19*9	0.	17.63	
WE(SO.IN)	• 0	6.61	• 0	17.63	•
INSPECTION INTERVALS (HRS)	ALS(HRS)				
INITIAL	25	500	1000	12000	
~	52	500	1250	15000	
•	<b>25</b>	500	1563	18750	
3	<b>52</b>	<b>5</b> 00	1953	23438	
\$	<b>52</b>	200	584	8203	
•	<b>5</b> 2	500	654	10254	
7	25	200	1068	12817	
αï	5≥	00 <b>2</b>	1335	16022	
•	ĸ	2n0	1669	20027	
NUMBER OF SPECIAL NUMBER OF STRUCTI FINAL ACTUAL AVER NUMBER OF AIRCRAF	NUMBER OF SPECIAL INSPECTIONS CONDUCTED: 1 NUMBER OF STRUCTURAL MODIFICATIONS: 0 FINAL ACTUAL AVERAGE WODIFIED FATIGUE LIFE: 547824 HOURS NUMBER OF AIRCRAFT MODIFIED IN SERVICE: 0	1 : 547824 HOURS 1			

STRUCTURAL FAILURES
AIRCRAFT YO. FLT. HUUNS

AIRCRAFT TYPE: MYBRID

60000 HDURS
AIRCHAFT SERVICE LIFE:
500
NUMMER OF AIRCHAFT IN FLEET:

	539763 HOURS
AF1-0090	ACTUAL AVERAGE FATIGUE LIFE: 539763 HOURS
STRUCTURAL ELEMENT: MSC-SMB-AFT-6090	PREDICTED AVERAGE FATIGUE LIFE: 593400 HOURS

FIRST CRACK CORROSIGN SERVICE DAMAGE  1603 5542 1603 48952 56707	.15	PRODUCTION DEFECTS	c	••••	• • • • • • • • • • • • • • • • • • • •
FIRST CRACK CORROSIGN  1603 5542  48952 58707	TION OF AIRCHAFT DEFEC	SERVICE DAMAGE	M	1603	48952
FIRST CRACK	AND TIME TO INITIAL	CORROSIGN	 •	5542	56707
	NUMBER	FIRST CRACK	 c	1603	48952

	AUMBER AND	LENGTH OF CHACKS DE	TECTED AT EACH LEVE	L UP INSPECTION	
	A-LEVEL	Bolevel	-LEVEL C-LEVEL D-LEVEL D-LEVEL	D-LEVEL	SPECIAL
	* * * * * * * * * * * * * * * * * * * *				
CCCURRENCES	Đ	~	~	**	0
E187183	-0	15.	67.	1.05	•
MAX (TX)	, ,	59.	.55	1.05	
AVG(IN)	, •	85.	.52	1.05	;

	A-LEVEL	H-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
	*****		****	* * * * * * *	
OLLURRENCES	0	n,	0	-	0
MIN(SD. IN)		1.65	•0	17.84	
MAKESO IN	Ċ	80.5	•0	17.84	•
AVG(SO.IN)	. 0	1.80	0	17.84	•
INSPECTION INTERVALSINAS)	VALS(HRS)				
TETTIA	ĸ	200	1000	12000	
	. X	002	1250	15000	
1 10		00≥	1563	18750	
1 4	500	002	1953	23438	
·	; <del>(</del>	002	484	6203	
• •	52	500	854	10254	
. ~	: £	200	1068	12817	
. «	×	200	1335	16022	
• •	<b>52</b>	002	1669	20027	

FINAL ACTUAL AVERAGE WODIFIED FATIGUE LIFE: 539763 HOURS NUMBER OF AIRCRAFT WODIFIED IN SERVICE: 0

STRUCTURAL FAILUMES
AIRCRAFT NO. FLT. HOURS

RESIDUAL STRENGTH EQUALS FAIL-SAFE STRENGTH AIRCRAFT NO. FLT. HOURS

- Continued Figure 5

AIRCHAFT TYPE: MYBRID

60000 HOURS		E: 976224 HOURS
AIRCRAFT SERVICE LIFE: 60000 HDURS	SC-S#8-AFT-0127	ACTUAL AVERAGE FATIGUE LIFE: 976224 HOURS
NUMGER OF AIRCRAFT IN FLEET: 500	STRUCTURAL ELEMENT: ASC-SAB-AFT+0127	PMEDICTED AVERAGE FATIGUE LIFE: 607200 HOURS

DEFECTS	
NUMBER AND TIME TO INITIATION OF AIRCHAFT DEFECTS	
ION OF A	
INITIAT	
TIME TO	
MBER AND	
3	
,	

	FIRST CRACK	CORROSION	SERVICE DAMAGE.	PRODUCTI	PRODUCTION DEFECTS
<b>OCCUPRE</b> VCES		m	0		¢
MIN(HRS)	46529	25520	0	:	
MAX (MKS)	46529	59353	c	:	
AVG(HRS)	46529	43145	Ø	<b>.</b>	•
	NUMBER AND LE	ENGTH OF CRACKS DET	NUMBER AND LENGTH OF CRACKS DETECTED AT EACH LEVEL OF INSPECTION	JF INSPECTION	
	A-LEVEL	8-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
		*****			1 1 1 1 1 1
OCCURRENCES	c	wel	0	0	0
HINGIN	•	3,33	•0	•0	•0
MAX(IN)	•	3,33	•0	•0	•
AVG(IN)	•	3.33	•0	•0	•

DCCURRENCES         1         1         0         1         1         0         2.96         0         2.96         0         2.96         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         0         2.96         0         2.96         0         2.96         0         2.96         0         0         2.96         0         2.96         0         2.96         0         0         2.96         0         0         2.96         0         2.96         0	INTERVALS(MRS)  INTERVALS(MRS)		A-LEVEL	B-LEVEL	C-LEVEL	D-LEVEL	SPECIAL
0	288 0		111111				
1.36 0. 2.96 0. 1.38 0. 2.96 0. 1.38 0. 2.96 0	38 0. 38 0. 30 0. 200 1000 200 1250 200 1563 200 1563 200 15684 200 1068	OCCURRENCES	6	-	0	شتو	0
1.38 0. 2.96  1.36 0. 2.96  INTERVALS(MRS)  25 200 1000 1200 25 200 1563 1550 25 200 1953 23438 25 200 884 10254 25 200 854 10254 25 200 1068 1250	38 0. 38 0. 200 200 200 200 200 200 200 2	M18(SQ. IN)	• 0	1.38	•0	2.96	•
INTERVALS(MRS)  25 200 1000 12000 255 200 1563 1855 2348 255 200 684 8203 2545 8203 255 200 854 10254 8201 1261 1261 1261 1261 1261 1261 1261 1	200 200 200 200 200 200 200 1955 200 1956 1958 200 1958	MAX (SQ. [M)	ç	1,38	•0	2.96	
INTERVALS(MRS)  25 200 1000 25 200 1250 25 200 1563 25 200 1953 25 200 884 25 200 854	200 200 200 200 200 200 200 1068 200 1068	AVG(SD.IN)		1.38	•0	2.96	• 0
25 200 1000 1250 200 1250 25 200 1250	200 200 200 200 200 200 200 354 200 1068	_	ERVALS (HRS)				
25 200 1250 200 255 20 200 200 200 200 200 200 2	200 1250 200 200 200 200 200 200 200 200 200	INITIAL	*2	902	1000	12000	
200 1563 200 1953 200 - 684 200 854 200 1068	200 1563 200 1953 200 854 200 1068	~	. 52	002	1250	15000	
200 1953 200 - 684 200 - 684 200 1068	200 1953 200 684 200 854 200 1068	~	\$2	0 <b>02</b>	1563	18750	
260 . 684 200 . 684 200 . 1068	200 - 684 200 - 684 200 - 1068	4	52	200	1953	23438	
200 A54 200 1068	200 854 200 1068	2	\$	560	789 .	B203	
200 106A	200 1068	•	<b>52</b>	200	854	10254	
		•	<b>\$</b> 2	002	1068	12817	

STRUCTURAL FAILUMES AIRCRAFT NO. FLI. HOLRS

Figure 5 - Continued

HESIDUAL STRENGTH EQUALS FAIL-SAFE STRENGTH AIRCRAFT NO. FLT. HOURS

CINCAR FACE LARGET

and a rest of two powers of the Park Andreas and

**************************************						CAECIAL.	ψ	<b>4 0</b>		<b>39</b> 6534.	35.67 35.87 35.87			EQUALS FAIL-SAFE STRENGTH FLT. HOUPS STA. NO.
riscossi Seavice Life:	<b>5</b> 72		<b>8</b>	•	DE BASKETTUR	D-1E VE	ब्रा ( <b>()</b> हिं हा	W W	EVEL OF INSPECTION	D-LEVEL	2.66 17.06 12.06	12000 8203 23458		STREET TO THE TREET THE TR
	Stauctukal Elévénts ascosadosfi	the estate	SEBNICE DAMAGE	1854 5945 59458 59458	EXEM LEVEL	CHESE	2 9	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	DETECTED AT EACH LEVEL	C-1EVEL	त <b>्र</b> ्	2 # M 2 # M 2 # # 8 # #		14002838 14002838
7 - SDP	Transport of Standings	and time to initiation	CCBROSION	13 5,575 5,675 5,565 5,565 5,665	TENGTH TO FALLES DETECTED AT	B-15 VEL	V 1	17 PM 14 17 PM 1 1 0 0 0	CORPOSION DEFECTS		2.0 V 2.0 V 2.0 V 3.0 V 3.0 V 3.0 V 3.0 V 3.0 V	999 982 7	CTED: 6 0 ICE: 9	STA, MO.
euragea de elegates la fleets	98 -	a desir.	FISSI CAREN	12 1653 50789 49795	at dus mademin	A-4.EVEL	ବ	4 4 1 4 9 00	THE PERSON CAME CONTRACTOR		ල ප <b>ිටි</b>	78 <u>1</u> .\$(HP\$) 25 25 25	SPECIAL INSPECTIONS CONDUCTED: STRUCTURAL MODIFICATIONS: 0 AIRCRAFT MIDIFIED IN SERVICE:	STAUCTURAL FAILURES FLI. MEURS
<b>可过在</b> 现行为				OCCURRENCES MINIMES) MAXIMES) AVS(MRS)			OCCURRENCES	等10位(10位) 医体统(10位) Actification			OCCURRENCES MIN(SO.IN) MAX(SO.IN) AVG(SO.IR)	IKSPECTION INTEGVALS(MRS) INITIAL SMGRTEST	NUMBER OF SPECIAL NUMBER OF STRUCTS NUMBER OF STRUCTS	STR ATRCRAFT WG,

Figure 5 - Concluded

The number of occurrences in the fleet and the times to initiation of the four types of aircraft defects considered by SAIFE are displayed next. Whenever there is a fatigue crack initiation and there are no other cracks in the element, a first crack is said to have occurred. A single element can experience more than one first crack in its lifetime by having a crack initiation after a repair. Similarly, there can be more than one occurrence of corrosion and service damage, although the program does not allow more than one corrosion or service damage defect to exist simultaneously in the same element.

Production defects are one-time occurrences unless there is a structural modification installed. These, too, can have production defects. The times to defect initiation are measured from the time when the aircraft enters service for the initial defects and from the time when the aircraft was last repaired for subsequent defects.

Noxt, the number and lengths of cracks detected in the fleet at each level of inspection are printed. These numbers include second and third crack detections. Following the crack detection output are the number and areas of corrosion defects detected in the fleet at each level of inspection. A history of the inspection interval changes is printed next. Each time that the aircraft service experience indicates that either an interval increase or an interval decrease is needed, the new interval values are printed. Although the number of interval changes allowed in the simulation is unlimited, the output array size limits the number printed to 30.

The number of floet-wide special inspections performed a printed next. Bach special inspection is always preceded by a decrease in inspection intervals. Defects detected during a special inspection can cause an additional decrease in inspection intervals. Next, the number of structural modifications developed is printed. This number includes modifications because of fatigue test failures or aircraft service experience. The final actual average fatigue life is printed next. If there have been no modifications, this number will be the same as that at the top of the page. If there have been modifications, this number is the actual average fatigue life of the most recently developed modification.

Shown next s the number of aircraft modified in service. If the only modification developed was due to a fatigue test failure, this number can be zero if the test life was such that retrofits were not required. If there were more than one modification requiring retrofits, this number can be greater than the size of the fleet. Finally, each time an aircraft experiences structural failure or its residual strength reaches its fail-safe strength, the aircraft number and the number of accrued flight hours are printed. The aircraft number is assigned by its relative time of entry into service. Aircraft No. 1 is the first aircraft to enter service.

### 3.2 Summary Data

The last section of Figure 5 illustrates a sample summary for the element type WSC-SWB-AFT. All the numbers represent a summary of all the specific elements of this type. Except for the shortest and longest inspection intervals, each number in the summary will appear in one of the specific element outputs. As indicated earlier, the number of interval changes allowed in the simulation is unlimited, and the number printed for a specific element is limited to 50. The shortest and longest intervals printed in the summary are determined from the unlimited number of changes occurring in the simulation.

# 3.3 Long List Data

Figure 6 illustrates a sample long list output for the input shown in Figure 4. For each element the long list headings contain the following: aircraft description, number of aircraft in the fleet, aircraft service life, element description, predicted average fatigue life of the element, actual average fatigue life of the element, and the initial inspection intervals. For each aircraft being tracked, the long list option causes selected information to be printed each time the program control passes to certain events and routines. These events and routines along with the information printed are the following:

Event ENTER, SERVICE - Prints aircraft identification number, number of hours from start of simulation, projected flight hours until crack initiations, and the slow and fast crack growth rates.

Routino INSTALL, MODIFICATION - Prints aircraft identification number, flight time on aircraft, flight hours until crack initiations, and the slow and fast crack growth rates.

Event IN. SERVICE, DAMAGE - Prints aircraft identification number and flight time on aircraft.

Event CORROSION - Prints aircraft identification number, flight time on aircraft, revised slow and fast crack growth rates, revised times until crack initiations, and revised time until failure.

livent 1.STRENGTH.REDUCTION - Prints aircraft identification number, flight time on aircraft, and projected flight hours until element failure.

Event 2.STREAGTH.REDUCTION - Prints aircraft identification number, flight time on aircraft, and projected flight hours until element failure.

dense samme de la company de la company

Event 3. STRENGTH, REDUCTION - Prints aircraft identification number, flight time on aircraft, and projected flight hours until element failure.

Event 1.ITE - Prints aircraft identification number, length of crack, and flight time on aircraft.

Event 2. ITE - Prints aircraft identification number, length of crack, and flight time on aircraft.

Event 3.1TE - Prints aircraft identification number, length of crack, and flight time on aircraft.

Event D.LEVEL.INSPECTION - Prints aircraft identification number and flight time on aircraft. If an inspection interval increase is implemented at this time, revised intervals are also printed.

Routine EXAMINE - For each defect found, prints size of defect, level of inspection, aircraft identification number, and flight time on aircraft.

Event REACH, FAIL, SAFE, LGT - Prints aircraft identification number and flight time on aircraft.

Event FAILURE - Prints aircraft identification number, flight time on aircraft, sum of crack lengths, and element residual strength.

Event RETIRE, FROM, SERVICE - Prints aircraft identification number and flight time on aircraft.

Event REPAIR - Prints aircraft identification number and projected times to crack initiations.

Event INCREASE, INSPECTION, FREQUENCY - Prints revised C-level and D-level inspection intervals.

Event IMMEDIATE. FLEET. INSPECTION - For each defect found, prints type of defect, size of defect, aircraft identification number, and flight time on aircraft.

### AIRCHAFT TYPE: HYRRID

NUMBER OF AIRCHAFT IN FLEET: 500

ATHCHAFT SERVICE LIFE: 60000 HOURS

STRUCTURAL FLEMENT: #8C-S#R-AFT-0000

PREDICTED AVERAGE FATIGUE LIFET TUSZOG HOURS

ACTUAL AVERAGE PATTGUE LIFFE 67010H MINIMS

INITIAL INSPECTION INTERVALS

A-LEVEL 75 HIURS H-LEVEL 200 HIURS C-LEVEL 1000 HIURS D-LEVEL 12000 HIURS

AZC NO. 100 ENTERS SERVICE STOO WOURS FROM START OF STAULATION

1ST CHACK INITIATION PHODECTED AT ASSSSE FLIGHT HOURS AND CHACK INITIATION PHODECTED AT ARREST FLIGHT HOURS ON CHACK INITIATION PHODECTED AT ASSOCIATION HOURS STONE FLIGHT HOURS STONE CHACK GROWTH WATER 2 .0000040 TECHES MOUNTERS AND CRACK GROWTH WATER 2 .001605 INCHES MOUNTERS AND CRACK GROWTH AND CRACK GR

THEREFORE THE TRITERVAL THERESE THREEMENTER CHIEVEL THEREVAL NOT 1250 HOURS HALFVEL THEREVAL HELF TERROR HOURS

AVE NO. FOR ENTERS SERVICE ISUSO HOURS FROM START OF STAULATTO

IST CRACK INTITATION PUBLICIED AT 788845 FLIGHT MOURS 200 CRACK INTITATION PROJECTED AT 110450 FLIGHT MOURS AND CRACK INTITATION PROJECTED AT 1312006 FLIGHT MOURS SUN CRACK GROWN MATE = .000074 IDDRESSMOUN FAST CRACK GROWN MATE = .001490 INCHESSMOUR

DELEVEL INSPECTION PERFORMED OF AND MO. 100 AT 12000 HOURS

AVE NOT SOM ENTERS SERVICE DENSO HOURS FROM START OF STAULATION

19T CHACK TRITIATION FROMFOTED AT RENION FLIGHT HOURS AND CHACK TRITIATION PROMETED AT RESPONDE FLIGHT HOURS SLOV CHACK TRITIATION PROMETED AT 12992A3 FLIGHT HOURS SLOV CHACK GRINTH RATE # .000094 INCHESZHOUR FAST CHACK GRINTH RATE # .0018A3 TECHESZHOUR

D-FEARE INSELLITUA BENEGRAFO OF TAC POP 500 91 15000 House

THISPECTION TRITERVAL THICKEASE THPLEMENTED CHILDRE THITRAVAL HILL TSEE HILLS OF THE STATE OF TH

DELEVEL 1 SPECITOR PERFORMEN OF AZO MO. 100 AT 27000 HOURS

DELEVEL INSPECTION PERFORMED OF AND FO. 300 AT 12000 HOURS

D-LEVEL 1-SOFCITOR DENEMBER OF AND NO. 200 AT 24600 HOURS

THISPECTION INTERVAL LICHEASE INPLEMENTED CHIEVEL INTERVAL GOVER 1953 HOURS DELEVEL THISPECAL GOVERNMENTS

DELEVEL THEORETING PERFERSED OF AZO FO. 166 AT 45750 HINES

D-LEVEL INSPECTION PERFORMED IN AZC CO. 300 AT 30750 HOURS

DELEVEL THEORETION PERFITHMED THE BYO HOLE POR BE 42750 HIGHS

AVE NOT TOO RETIRED FROM SERVICE AT ADDOD FLIGHT HOURS

AZC NO. 200 RETIMEN FROM SERVICE AT BORRO FLIGHT HOURS

DELEVEL TUSPECTION PERFORMED ON AZO NO. 300 AT SULAR HOURS

AZC NO. 300 RETIRED SHOW SERVICE AT ADDOD FLIGHT HOURS

THISPECTION IN FRVAL DECREASE IMPLEMENTED CALEVEL 1 TERRVAL MINE AND HOURS DALEVEL THISPERVAL MINE 8203 HOURS

a. Structural Element: WSC-SWB-AFT-0000

Figure 6. Sample Long List Output Produced by Input Shown in Figure 4

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LURANG LIVENAME TO CHECKE MINING TO SEE MINING UNITED TO CHECKET TO LEAN AT THE MINING TO CHECKE THE TOTAL TO CHECKE THE TOTAL THE MENT TO CHECKE THE TOTAL THE MENT TO CHECKET THE MINING TO CHECKET THE MENT THE MENT TO CHECKET THE MENT THE MENT THE MENT THE MENT THE MENT THE MENT THE M

[NGPECTION INTERVAL INCHEASE INDITIONS OF THE TOTAL PROPERTY INTERVAL NO. 12817 HOURS

a. Structural Element: WSC-SWB-AFT-0000 (Concluded)

AIRCRAFT TYPE: HYBRID

NUMBER OF AIRCRAFT IN FLEETS 500

ATRORAFT SERVICE LIFE: MODOR HOURS

STRUCTURAL ELEMENT: WSC-SWR-AFT-0040

PHEDICTED AVERAGE FATIGUE LIFF'S 690000 HOURS

ACTUAL AVERAGE FATIGUE LIFF! 424261 HOURS

INITIAL INSPECTION INTERVALS

A-LEVEL 25 HOURS H-LEVEL 200 HOURS C-LEVEL 1000 HOURS D-LEVEL 12000 HOURS

AZC NO. AN ENTERS SERVICE - 4400 HOURS FROM START OF SIMULATION

IST CRACK INITIATION PROJECTED AT STUDBO FLIGHT HOURS AND CRACK INITIATION PROJECTED AT 644584 FLIGHT HOURS FOR CRACK INITIATION PROJECTED AT 691441 FLIGHT HOURS SLOW CRACK GROWTH RATE = .000073 INCHES/HOUR FAST CRACK GROWTH HATE # .001468 INCHES/HOUR

INSPECTION INTERVAL INCREASE IMPLEMENTED C+LEVEL INTERVAL NOW 1250 HOURS D+LEVEL INTERVAL NOW 15000 HOURS

D-LEVEL INSPECTION PERFORMED ON AZO NO. BK AT 12000 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED
C-LEVEL INTERVAL NOW 1563 HOURS
D-LEVEL INTERVAL NOW 18750 HOURS

D-LEVEL INSPECTION PERFORMED ON AIC MO. 86 AT 27000 HOURS

AZC NO. 497 ENTERS SERVICE 44750 HOURS FROM START OF SIMULATION

IST CRACK INITIATION PROJECTED AT SIMMIM FLIGHT HOURS AND CRACK INITIATION PROJECTED AT SAMBAM FLIGHT HOURS AND CRACK INITIATION PROJECTED AT 658MM FLIGHT HOURS SLOW CRACK GROWTH RATE = .000073 INCHES/HOUP FAST CRACK GROWTH RATE = .001468 INCHES/HOUR

INSPECTION INTERVAL INCREASE IMPLEMENTED
C -LEVEL INTERVAL NUM 1953 HOURS
D-LEVEL INTERVAL NOW 23438 HOURS

D-LEVEL INSPECTION PERFORMED ON A/C NO. 86 AT 45750 HOURS
D-LEVEL INSPECTION PERFORMED ON A/C NO. 497 AT 12000 HOURS
A/C NO. 86 RETIRED FROM SERVICE AT 60000 FLIGHT HOURS
D-LEVEL INSPECTION PERFORMED ON A/C NO. 497 AT 35438 HOURS
D-LEVEL INSPECTION PERFORMED ON A/C NO. 497 AT 58875 HOURS
A/C NO. 497 RETIRED FROM SERVICE AT 60000 FLIGHT HOURS

b. Structural Element: WSC-SWB-AFT-0030 Figure 6 - Continued

### AIRCHAFT TYPE: HYRRID

NUMBER OF AIRCRAFT IN FLEET: 500

AIRCRAFT SERVICE LIFE: AGGOD HOURS

STRUCTURAL ELEMENT: WSC-SWH-AFT-0060

PREDICTED AVERAGE FATIGUE LIFE: 662400 HOURS

ACTUAL AVERAGE FATTGUF LIFFT SUTHAU HOURS

INITIAL INSPECTION INTERVALS

A-LEVEL 25 HOURS H-LEVEL 200 HOURS C-LEVEL 1000 HOURS D-LEVEL 12000 HOURS

A/C NO. 1 FRIERS SERVICE 150 HOURS FROM START OF SIMULATION

IST CRACK INITIATION PROJECTED AT 37922 FLIGHT HOURS 2ND CRACK INITIATION PHOJECTED AT 1099412 FLIGHT HOURS 3RD CRACK INITIATION PROJECTED AT 11588H3 FLIGHT HOURS SLOW CRACK GROWTH RATE # .000089 INCHES/HOUR FAST CRACK GROWTH RATE # .00177# INCHES/HOUR

D-LEVEL TYSPECTION PERFORMED ON AZE NO. 1 AT 12000 HOURS

INSPECTION INTERVAL INCREASE INPLEMENTED
CHLEVEL INTERVAL NOW 1250 HOURS
DHLEVEL INTERVAL NOW 15000 HOURS

D-LEVEL INSPECTION PERFORMED ON AZT NO. | LAT 24000 HOURS

INSPECTION INTERVAL INCHEASE IMPLEMENTED C-LEVEL INTERVAL NOW 1563 HOURS D-LEVEL INTERVAL NOW 18750 HOURS

D-LEVEL THEPECTION PERFORMED OF AIR MO. I AT \$9000 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED C-LEVEL INTERVAL NUM 1953 HOURS D-LEVEL INTERVAL NOW 23438 HOURS

INSPECTION INTERVAL OF CHEASE IMPLEMENTED C-LEVEL INTERVAL NOW 684 HOURS D-LEVEL INTERVAL NOW 8203 HOURS

FLEET WIDE SPECIAL INSPECTION PERFORMED

D-LEVEL INSPECTION PERFORMED ON AZO NO. | AT 53806 HOURS

INSPECTION INTERVAL INCHEASE IMPLEMENTED C-LEVEL INTERVAL NOW 854 HOURS D-LEVEL INTERVAL NOW 10254 HOURS

A/C NO. 1 RETIRED FROM SERVICE AT ACCOUNTINGHT HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED C-LEVEL INTERVAL NOW 1068 HOURS D-LEVEL INTERVAL NOW 12817 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED C-LEVEL INTERVAL NOW 1335 HOURS D-LEVEL INTERVAL NOW 16022 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED C-LEVEL INTERVAL NOW 1669 MOURS D-LEVEL INTERVAL NOW 20027 HOURS

c. Structural Element: WSC-SWB-AFT-0060

Figure 6 - Continued

AIRCRAFT TYPE: MYHRID ATRCHAFT SERVICE LIFEL - HOODS HOURS NUMBER OF ATRCHAFT IN FLEETE 500 STRUCTURAL PLEMENTS 48C+S+B-AFT+0040 ACTUAL AVERAGE PATTICUE LIFE: SEGINE HILLIPS PREDICTED AVERAGE FATIGUE LIFE: 543400 HOURS INITIAL INSPECTION INTERVALS SHOON OUS H-LEVEL 1000 HOURS ROLEVEL 12000 HOURS 550 HOURS FHOM START OF SIMULATION AJC NO. 9 ENTERS SERVICE IST CRACK INTITATION PROJECTED AT 479151 FLIGHT MOURS PND CRACK TWITTATION PROJECTED AT 1077598 FLIGHT MOURS JND CRACK INITIATION PROJECTED AT 1249581 FLIGHT MOURS SLOW CRACK GROWTH WATE # . OODNAT INCHESPHOUR FAST CRACK GROWTH WATE # . OF THE THOUGHT KOO HOURS FROM START OF SIMULATION AZC NIL EN ENTERS SERVICE IST CHACK INTITATION PROJECTED AT ASHORR FLIGHT MOUNS AND CRACK INTITATION PROJECTED AT TOHOGY FLIGHT MOUNS THE CRACK INTITATION PROJECTED AT RHUODS FLIGHT HOURS SLOW CRACK GROWTH HATE & .000074 INCHESTMOUR FAST CRACK GROWTH RATE & .0014AS INCHESTMOUR ASO HOURS FROM START OF SIMULATION AVC NO. 11 ENTERS SERVICE IST CRACK INITIATION PROJECTED AT 370200 FLIGHT HOURS AND CRACK INITIATION PROJECTED AT ARANJ FLIGHT HOURS AND CRACK INITIATION PROJECTED AT 114517 FLIGHT HOURS SLOW CHECK CHUNTH HATE & TOUGHA THUMENTHOUR PART CHECK CHUNTH HATE & TOUGHAS INCHESTHOUR 700 HOURS FRUM START OF STRUCATION AVC NO. 12 ENTERS SERVICE

IST CRACK INITIATION PHOJECTED AT 64424 FLIGHT HOURS AND CRACK INITIATION PHOJECTED AT 651954 FLIGHT HOURS AND CRACK INITIATION PHOJECTED AT 724727 FLIGHT HOURS SLOW CRACK ORDER RATE # .000014 INCHES/HOUR FAST CRACK GROWTH RATE # .001476 INCHES/HOUR DELEVEL INSPECTION PERFORMED ON A/C NO. 9 AT 12000 HOURS

DILEVEL INSPECTION HEREORNED ON AVE NO. TO AT 12000 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED CHLEVEL INTERVAL NOW 15000 HOURS OMLEVEL INTERVAL NOW 15000 HOURS

the Statement of the

D-LEVEL INSPECTION PERFORMED ON AZO NO. 11 AT 12000 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 12 AT 12000 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 9 AT 24000 HOURS

D-LEVEL INSPECTION PERFORMED ON AIC NO. 10 AT 21000 HOURS

INSPECTION INTERVAL INCHEASE IMPLEMENTED C-LEVEL INTERVAL NOW 1563 HOURS D-LEVEL INTERVAL NOW 14750 HOURS

DELEVEL INSPECTION PERFORMED ON AVE NO. 11 AT 27000 HOURS DELEVEL INSPECTION PERFORMED ON AVE NO. 12 AT 39000 HOURS DELEVEL INSPECTION PERFORMED ON AVE NO. 9 AT 39000 HOURS DELEVEL INSPECTION PERFORMED ON AVE NO. 10 AT 05750 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED

C-LEVEL INTERVAL NOW 1953 HOURS

D-LEVEL INTERVAL NOW 23418 HOURS

d. Structural Element: WSC-SWB-AFT-0090

Figure 6 - Continued

Delevel Interval interval interval interval interval interval interval interval interval of and and and in a stand minura interval interval of an and and in a stand minura interval interval of an and and in a stand minura interval interval of and and and in a stand minura interval interval interval of and and in a stand minura interval interval independent of and in a stand minura interval interval independent interval independent interval independent interval independent interval independent interval independent interval interval independent interval independent interval independent interval interval independent ind

AVC NO. . 9 RETIRED FROM SERVICE AT ENGOD FLIGHT MOUNS

AVE NO. TO RETIMED FROM BENVICE AT MODOR FLIGHT MOURS

AZC NO. IN RETIRED FROM SERVICE AT COORD FLIGHT HOURS

AVO NO. 13 RETTRED ERON SERVICE AT EDUNG FLIGHT HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED CHLEVEL INTERVAL NOW 10AM HOURS OHLEVEL INTERVAL NOW 12M17 HOURS

INSPECTION INTERVAL INCHEASE IMPLEMENTED CHLEVEL INTERVAL NOW 1335 HOURS OFLEVEL INTERVAL NOW IMOR? HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED COLEVEL INTERVAL NOW 1669 HOURS OFLEVEL INTERVAL NOW 20027 HOURS

d. Structural Element: WSC-SWB-AFT-0090 (Concluded)

AIRCHAFT TYPET HYHRID

WINNER OF AIRCRAFT IN PLEFTE 500

ATRUNANT SERVICE LIFE: MODOD HOURS

STRUCTURAL ELEMENTS WSC-SAR-MET-0127

PREDICTED AVERAGE FATIGUE LIFET 607700 HOURS

ACTUAL AVERAGE FATIGUE LIFFE STARRE HOURS

INITIAL INSPECTION INTERVALS

A=LFVEL 25 HOURS
H=LEVEL 2000 HOURS
C=LEVEL 12000 HOURS

INSPECTION INTERVAL INCHEASE IMPLEMENTED COLEVEL INTERVAL NOA 1250 HOURS O-LEVEL INTERVAL NOA 15000 HOURS

AVC NO. 323 ENTERS SERVICE 27350 HOURS FRUM START OF STYULATION

IST CRACK INITIATION PROJECTED AT 115105P FLIGHT HOURS SND CRACK INITIATION PROJECTED AT 1562644 FLIGHT HOURS SND CRACK INITIATION PROJECTED AT 178357H FLIGHT HOURS SUCH CRACK GROWTH RATE \$ .000074 INCHES/HOUR FAST CRACK GROWTH RATE \$ .001487 INCHES/HOUR

INSPECTION INTERVAL INCREASE IMPLEMENTED
C-LEVEL INTERVAL NOW 1563 HOURS
D-LEVEL INTERVAL NOW 18750 HOURS

e. Structural Element: WSC-SWB-AFT-0127

Figure 6 - Continued

O-LEVEL INSPECTION PERFORMED ON AZC NO. 123 AT 12000 HOURS

AZC NO. 456 ENTERS SERVILE 40650 HOURS FROM START OF SIMULATION

1ST CRACK INITIATION PROJECTED AT 488494 FLIGHT HOURS
250 CRACK INITIATION PROJECTED AT 1391054 FLIGHT HOURS
180 CRACK INITIATION PROJECTED AT 1602495 FLIGHT HOURS
180 CRACK GROWTH RATE = .000095 INCHES/HOUR

AZC NO. 472 ENTERS SERVICE 42250 HOURS FROM START OF SIMULATION

1ST CRACK INITIATION PROJECTED AT 1436062 FLIGHT HOURS
250 CRACK INITIATION PROJECTED AT 1636062 FLIGHT HOURS
350 CRACK INITIATION PROJECTED AT 2704671 FLIGHT HOURS
350 CRACK INITIATION PROJECTED AT 2704671 FLIGHT HOURS
151 CRACK GROWTH RATE = .000066 INCHES/HOUR

108PECTION INTERVAL NOW 1453 HOURS

D-LEVEL INTERVAL NOW 1453 HOURS

D-LEVEL INTERVAL NOW 2543H HOURS

D-LEVEL INTERCTION PERFORMED ON AZC NO. 472 AT 12000 HOURS

LOST TO THE PROTECTION PERFORMED ON AZC NO. 472 AT 12000 HOURS

LOST THEREFORE THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMED ON AZC NO. 473 AT 12000 HOURS

LOST THE PERFORMENTED

C-LEVEL INTERVAL NOW BAY HOURS D-LEVEL INTERVAL NOW HEOS HOURS FLEET HIDE SPECIAL INSPECTION PERFORMEN

White the second section is

INSPECTION INTERVAL INCHEASE IMPLEMENTED C-LEVEL INTERVAL NOW 854 HOURS DELEVEL INTERVAL NOW 10254 HOURS

D-LEVEL INSPECTION PERFORMED ON AZO NO. 325 AT 45041 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 456 AT 31741 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 472 AT 30141 HOURS

INSPECTION INTERVAL INCREASE IMPLEMENTED
CHLEVEL INTERVAL NON 1068 HOURS
DHLEVEL INTERVAL NON 12817 HOURS

D-LEVEL INSPECTION PERFORMED ON AZO NO. 123 AT 55295 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 456 AT 41995 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 472 AT 40395 HOURS AZO NO. 323 RETIRED FROM SERVICE AT 60000 FLIGHT HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 456 AT 54812 HOURS D-LEVEL INSPECTION PERFORMED ON AZO NO. 472 AT 53212 HOURS AZO NO. 456 RETIRED FROM SERVICE AT 60000 FLIGHT HOURS

e. Structural Element: WSC-SWB-AFT-0127 (Concluded)

Figure 6 - Concluded

### APPENDIX A

# DETAILED PROGRAM DESCRIPTION

In the following detailed description of the SAIFE program, each event and routine in the program is presented separately. Except for the PREAMBLE, each presentation consists of a description, the definition of the local variables, if any, and a flow chart to illustrate the logic of the event or routine.

# APPENDIX A

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### 1. PREAMBLE

The PREAMBLE is the definition section of a SIMSCRIPT program. All global variables and global arrays are defined. Temporary entities are defined and tally statistics are identified. Event notices and functions are defined and an event priority order is set. The global real variables are listed and described below in the order in which they appear in the PREAMBLE. Global real variables which are input variables are not included here but can be found in the input section.

#### Global Real Variables

FLEET.STR.RED - This variable is the sum of crack lengths found in the fleet since the last inspection frequency change.

<u>1AAFL</u> - This variable is the actual average fatigue life of the element design determined in the MAIN program.

CRRF - Assigned a value in routine INITIALIZATION according to the element corrosion resistance rating, this variable is multiplied by the aircraft corrosion growth rate to give the element corrosion growth rate.

COST.OF. REPAIRS - This variable is the sum of repair costs for the fleet since the last modification.

FIXIT.COST - This variable is the cost of repairing a defect found at a particular inspection level. Its value is set in the inspection events.

CHG.FREQ.TIME - This variable is set equal to TIME.V whenever an inspection interval change is scheduled in the event REPAIR.

<u>1CRKT</u> - Each time a first crack occurs, this variable is set equal to the service time on the aircraft.

1CORT - Each time corrosion occurs, this variable is set equal to the service time on the aircraft.

<u>ISDT</u> - Each time service damage occurs, this variable is set equal to the service time on the aircraft.

ACRKL, BCRKL, CCRKL, DCRKL, SCRKL - Each time a crack is found during an A-level, B-level, C-level, D-level, or Special inspection, the corresponding variable is set equal to the crack length.

ACA, BCA, CCA, DCA, SCA - Each time corrosion is found during an A-level, B-level, C-level, D-level, or Special inspection, the corresponding variable is set equal to the corrosion area.

AIRFRAME.TIME - This variable is the number of flight hours accumulated since the last modification for aircraft no longer in service.

GICRK - Each time a first crack occurs, this variable is set equal to the service time on the aircraft.

G1COR - Each time corrosion occurs, this variable is set equal to the service time on the aircraft.

GISD - Each time service damage occurs, this variable is set equal to the service time on the aircraft.

GACRK, GBCRK, GCCRK, GDCRK, GSCRK - Each time a crack is found during an A-level, B-level, C-level, D-level, or Special inspection, the corresponding variable is set equal to the crack length.

GACA, GBCA, GCCA, GDCA, GSCA - Each time corrosion is found during an A-level, B-level, C-level, D-level, or Special inspection, the corresponding variable is set equal to the corrosion area.

CINSL, DINSL - Each time there is an inspection interval change, these variables are set equal to the C-level and D-level intervals, respectively.

KSMP - This variable is set equal to 1.0 in the A-level, B-level, and C-level inspection events and set equal to the D-level sampling percentage in the D-level inspection event.

The global integer variables are listed next. Again, input variables are not included in this list.

### Global Integer Variables

II) - In each event and routine, this variable is the identification number of the aircraft being processed.

IDCK - This variable is initialized to zero and incremented by one each time an aircraft enters service.

I - This variable is used as a local index or array subscript in different locations in the program.

COUNT.ELEMENT - Each time new element data is read in, this variable is incremented by one.

NICHG - This variable is the number of times that the inspection intervals have changed.

LHTA - This variable is the identification number of the aircraft among the ten high-time aircraft with the fewest flight hours.

And the second second with the second se

- 1.00M, OF, RETERM: the variable is the number of alreraft that have been retired from service.
- 2, NUM, OF, GRASH. This variable is the number of aircraft which have been removed from service because of structural failure,
- TRNL: This variable is the numeric identification of the lowest internal level of inspection.
- LYT, INSP, LAVID, This variable is the numeric identification of the lowest external level of inspection.
- Ill, the These variables are the numeric identifications of the lowest internal and external levels of inspection, respectively. If either of these variables is less than three, it is set equal to three.
- TO, M., MODIFIED This variable is the number of aircraft with a pending retrofit modification.
- MAN, NODIFIAD This variable is the number of aircraft that have had a current retrofit modification installed.
- thick this variable is the number of aircraft in service when a modification is implemented because of a fatigue test failure.
- OICR, OCOR, OSDM, OPD Those variables are the number of occurrences of first cracks, corrosion, service damage, and production defects, respectively, for a particular element.
- OSCR, OSCO Those variables are the number of cracks and corresion defects, respectively, detected during a special inspection for a particular element.
- NSIC This variable is the number of special inspections conducted for a particular element.
- NSMD This variable is the number of aircraft modified in service for a particular element.
- NSFL This variable is the number of aircraft experiencing structural failure for a particular element.
- NMD This variable is the number of structural modifications made on a particular element.
- NRFS This variable is the number of aircraft with the residual strength for a particular element reaching the fail-safe strength.

A STATE OF THE PROPERTY OF THE

- $88Ri8 \pm 1618$  variable is the number of aircraft with the residual strength for a particular element type reaching the fail-safe strength.
- d: this variable is used as a local index or array subscript in different locations in the program.
- LDA: If the long list option is in effect, this variable is the ascending sequential position of the element being processed among those elements read in under the long list option.
- GOICE, GOCOR, GOSOM These variables are the number of occurrences of Tirst cracks, corresion damage, and service damage, respectively, for a particular element type.
- GOSCR, GOSCO: These variables are the number of cracks and corrosion defects, respectively, detected during a special inspection for a particular element type.
- GOPO This variable is the number of occurrences of production defects for a particular element type.
- SNSIC this variable is the number of special inspections conducted for a particular element type.
- <u>SNMD</u> This variable is the number of structural modifications made on a particular element type,
- SNSMD This variable is the number of aircraft modified in service for a particular element type.
- SNSFL This variable is the number of aircraft experiencing structural failure for a particular element type.

The real arrays are listed next. Unless otherwise noted, all arrays are 1-dimensional. As before, input arrays are not included in this list.

#### Real Arrays

-

- C.INTERVAL, D.INTERVAL . The elements of these arrays are the current C-level and D-level inspection intervals for each aircraft in the fleet.
- ABCD This array is of size four and contains the most recent intervals for each of the four levels of inspection.
- CKREP. Time This array is the simulation time of the most recent crack repair for each aircraft.

CORRECTIME: This array is the simulation time of the most recent correston repair for each aircraft.

1.AST, SD = This array is the simulation time of the most recent occurrence of service damage for each aircraft.

occur, NOD = This array is the simulation time when the most recent modification was installed for each aircraft.

MSR, MFR : These arrays are the slow and fast crack growth rates, respectively, for each aircraft.

80,80. These arrays contain each of the inspection interval changes for the C-level and D-level, respectively.

NRN - This array is the random number selected to calculate the time until structural failure for each aircraft.

MRDD - This array is the simulation time of the most recently detected defect at either a C-level or N-level inspection for each of the ten high-time aircraft.

GCR1 - This array is the corresion multiplying factor for each alreadt.

The following are the integer arrays. Again, unless otherwise noted, all arrays are 1-dimensional and input arrays are not included.

### Integer Arrays

是现代是对对语言,也是是一种对象自然是是否是对对的的理解的自然的的。 第一 AISR, A2SR, A3SR - These arrays contain the event notice identification numbers for each aircraft for the events 1.STRENGTH, REDUCTION, 2.STRENGTH.REDUCTION, and 3.STRENGTH.REDUCTION, respectively.

 $\Delta E$  - This array is the event notice identification number for each aircraft for event FAILURE.

AIRPLANE - This array is the temporary entity identification number for each aircraft.

AAL, ABL, ACL, ADL - These arrays are the event notice identification numbers for each aircraft for the events A.LEVEL.IN-SPECTION, B.LEVEL.INSPECTION, C.LEVEL.INSPECTION, and D.LEVEL.INSPECTION, respectively.

AC, ATII - These arrays are the event notice identification numbers for each aircraft for events COROSION and T.INSPECTION. INCREASE, respectively.

- $\frac{ACID}{alreraft}$  = This array contains the identification numbers of those alreraft experiencing structural failure for a particular element.
- OICR, OICO These arrays are the number of cracks and corresion defects, respectively, detected at each of the four levels of inspection for a particular element.
- SACID This array contains the identification numbers of those alreraft experiencing structural failure for a particular element type.
- GOICE, GOICO These arrays are the number of cracks and corresion defects, respectively, detected at each of the four levels of inspection for a particular element type.
- III. TIME, ACRET This array contains the identification numbers of the ten high-time aircraft.
- APID This array contains the identification numbers of those alreaft with a particular element whose residual strength has reached the fail-safe strength.
- $\frac{\mathrm{SAP1D}}{\mathrm{alrer}}$  This array contains the identification numbers of those alreraft with a particular element type whose residual strength has reached the fail-safe strength.
- STIM This array contains the flight hours on each aircraft when the residual strength for a particular element reaches the fail-safe strength.
- SSTIM This array contains the flight hours on each aircraft when the residual strength for a particular element type reaches the fail-safe strength.
- FLTHR This array contains the flight hours on each aircraft when structural failure occurs for a particular element.
- SFLTHR This array contains the flight hours on each aircraft when structural failure occurs for a particular element type.
- ARFSL This array is the event notice identification number for each aircraft for event REACH.FAIL.SAFE.LGT.
- AlE, A2E, A3E These arrays are the event notice identification numbers for each aircraft for events 1.ITE, 2.ITE, and 3.ITE, respectively.

The global alpha arrays are listed next. As before, input arrays are not included in this list.

### Global Alpha Arrays

- 1.CR.EXISTS, 2.CR.EXISTS, 3.CR.EXISTS The elements of these arrays are set equal to "YES" for each aircraft whenever there is a first, second, and third crack initiation, respectively.
- CO.EXISTS This array is set equal to "YES" for each aircraft when it has corrosion initiation.
- SD. SCH This array is set equal to "YES" for each aircraft that has event IN. SERVICE. DAMAGE scheduled.
- SSTAN This array is the station number which identifies each aircraft experiencing the failure of a particular element type.
- SELNB This array is the station number which identifies each alreaft with a particular element type whose residual strength has reached the fail-safe strength.
- AIL, FSH This array is set equal to "YES" for each aircraft when vents REACH. FAIL. SAFE. LGT and FAILURE, respectively, are scheduled.
- IE1, IE2, IE3 This array is set equal to "YES" for each aircraft that has events 1.ITE, 2.ITE, and 3.ITE, respectively, scheduled.
- TMOD. PENDING This array is set equal to "YES" for each aircraft that has a modification pending because of a fatigue test failure.
- SMOD. PENDING This array is set equal to "YES" for each aircraft that has a modification pending because of service experience.
- INSP.SCH This array is set equal to "YES" for each aircraft that has inspections below the overhaul level scheduled.
- 1.INT, 2.INT, 3.INT These arrays are set equal to "YES" for each aircraft that has a first crack, second crack, or third crack, respectively, initiated internally.
- C.INT This array is set equal to "YES" for each aircraft that has corrosion initiated internally.

The temporary entity definitions and tally statements are self-explanatory. The events, functions, and routines are described in detail in the following sections.

### 2. MAIN

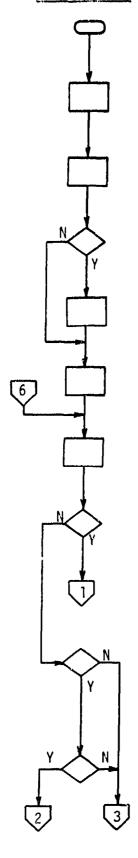
## 2.1 Description

In the MAIN program, space is reserved for all global arrays. The following operations are performed in the order given: all input data is read in; the actual average fatigue life of the element type is calculated; the necessity of a structural modification because of a fatigue test failure is determined; the first event ENTER. SERVICE is scheduled; and the simulation is initiated.

### 2.2 Local Variables

NFTS - This real variable is the time in flight hours from when the second production rate goes into effect to when the last aircraft enters service.

<u>SATL</u> - This real variable is the earliest simulation time at which a structural modification because of a fatigue test failure is ready for installation.



MAIN.

Reserve aircraft arrays.

Read aircraft data.

Does LONG.LIST="YES"?

Read in elements and aircraft to be tracked.

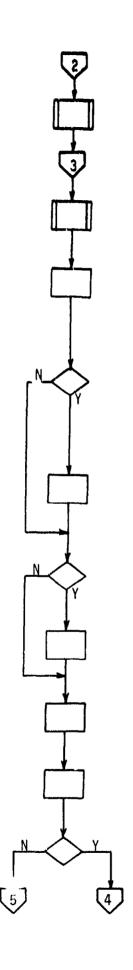
Reserve element arrays.

Read element data.

Does ELEMENT(1)="EOD"?

Is this element a different element type than previous element?

Does LONG.LIST="NO"?



Call routine SUMMARY.

Call routine INITIALIZATION.

Calculate actual average fatigue life of element type.

Does fatigue test failure occur at less than twice the aircraft service life?

Schedule a structural modification.

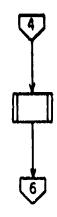
Is long list option in effect?

Print long list headings.

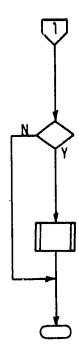
Schedule event ENTER.SERVICE.

Start simulation.

Does LONG.LIST="NO"?



# Call routine DISPLAY. OUTPUT.



Does LONG.LIST="NO"?

Call routine SUMMARY.

END.

# 3. INITIALIZATION

# 3.1 Description

This routine is called immediately after reading each new set of element input data. This routine changes the inspection level codes to numeric values, sets the corrosion growth multiplying factor based on the corrosion resistance rating, and resets the tally counters. It also initializes all the element global variables which are not part of the input. This routine is called from the MAIN program.

# 3.2 Local Variables

There are no local variables in this routine.

## 3.3 Flow Chart

7	Routine INITIALIZATION.
	Change inspection level codes to numeric values.
	Set corrosion growth multiplying factor.
	Reset tally counters.
	Initialize global variables.
	Return.

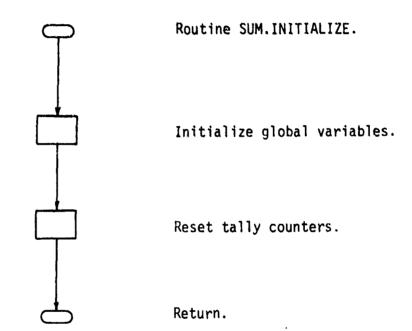
# 4. SUM. INITIALIZE

# 4.1 Description

This routine is called each time a new element type is read in. The element type is identified by the first twelve characters of the element identification. This routine initializes the global variables and resets the tally counters. This routine is called from the MAIN program.

### 4.2 Local Variables

There are no local variables in this routine.



#### 5. REAL.LIFE

### 5.1 Description

This routine accepts (1) the predicted average fatigue life of a particular element design and (2) the mean and standard deviation of the log-normal distribution of the ratio of the actual average fatigue life to the predicted average fatigue life. A random selection is made from the distribution and multiplied by the predicted average fatigue life. The resulting actual average fatigue life is returned to the calling routine. REAL.LIFE can be called from the MAIN program and events IMPLE-MENT.MODIFICATION and T.IMPLEMENT.MOD.

### 5.2 Local Variables

MEAN - This real variable, whose value is passed from the calling routine, is the mean of the ratio distribution.

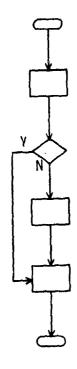
RATIO - This real variable, determined to be log-normally distributed, is the ratio of the actual fatigue life of an element design to its predicted fatigue life.

STD.DEV - This real variable, whose value is passed from the calling routine, is the standard deviation of the ratio distribution.

PDL - This real variable is the design predicted average fatigue life passed from the calling routine.

RFL - This real variable is the element actual average fatigue life which is returned to the calling routine.

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Routine REAL.LIFE.

Randomly select RATIO from distribution.

Is RATIO within allowable limits?

Set RATIO to limit.

Calculate actual average fatigue life.

Return with actual average fatigue life.

### 6. ENTER.SERVICE

### 6.1 Description

This event represents the entry into service of a new aircraft. The temporary entity AIRCRAFT is created and identified by the variable AIRPLANE(ID). The entity attributes TAIL.ID and ENTRY.TIME are defined and the AIRPLANE is filed in the set ACTIVE.FLEET. The routine FATIGUE.LIFE.SCATTER is called to determine the times to first, second, and third crack initiations. The slow and fast crack growth rates are calculated. The times to corrosion initiation and service damage are calculated. If either of these times is less than the service life of the aircraft, the corresponding defect is scheduled. If there is a production defect, the time to first crack initiation is replaced by a time drawn from a distribution of times to crack initiation of aircraft with production defects. If the long list option is in effect for each aircraft being tracked, this routine prints the following: (1) aircraft identification and time it enters service, (2) times to crack initiations, and (3) slow and fast crack growth rates. Crack initiations, D-level inspection, and retirement from service are also scheduled. If the present aircraft is not the last aircraft of the fleet, another ENTER. SERVICE is scheduled. This event can only be scheduled in the MAIN program and within itself.

### 6.2 Local Variables

DEFECT.LIFE - This real variable is the time to first crack initiation when the aircraft has a production defect.

HOURS. TO. CORROSION - This real variable is the time to corrosion initiation.

SECOND.LIFE - This real variable is the time to second crack initiation.

STD.SLOW - This real variable is the standard deviation of the distribution of slow crack growth rates.

FIRST.LIFE - This real variable is the time to first crack initiation when the aircraft has no production defect.

OURS.TO. SERVICE. DAMAGE - This real variable is the time to service damage occurrence.

RN - This real variable is a uniformly distributed random number between zero and one.

STD.FAST - This real variable is the standard deviation of the distribution of fast crack growth rates.

THIRD.LIFE - This real variable is the time to third crack initiation.

Event ENTER SERVICE.

Create an AIRCRAFT and file in ACTIVE.FLEET.

Determine times to first three crack initiations.

Calculate slow and fast crack growth rates.

Determine times to corrosion initiation and service damage.

Is time to corrosion initiation less than aircraft service life?

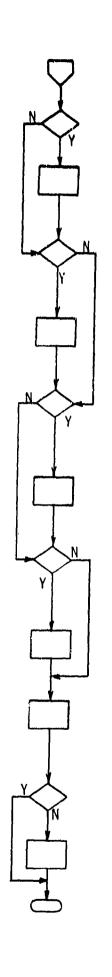
Schedule a corrosion initiation.

Is time to service damage less than aircraft service life?

Schedule a service damage occurrence.

Is random number less than probability of a production defect?

Determine new time to first crack initiation.



Is long list option in effect?

Print long list output if aircraft is one being tracked.

Is time to first crack initiation less than aircraft service life or is corrosion initiation scheduled?

Schedule first crack initiation.

Is time to second crack initiation less than aircraft service life or is corrosion initiation scheduled?

Schedule second crack initiation.

Is time to third crack initiation less than aircraft service life or is corrosion initiation scheduled?

Schedule third crack initiation.

Schedule first D-level inspection and aircraft retirement.

Is this last aircraft to enter service?

Schedule event ENTER.SERVICE.

Return.

# 7. FATIGUE.LIFE.SCATTER

### 7.1 Description

This routine receives the actual average fatigue life of the element design from the calling routine and returns the times to crack initiation of the first three fatigue cracks for the element in a particular aircraft. These times are random selections from a two-parameter Weibull distribution. This routine can be called from routine INSTALL.MODIFICATION and events ENTER.SERVICE and REPAIR.

### 7.2 Local Variables

ALPHA - This real variable is the shape parameter of the fatigue life distribution.

FIRST.LIFE - This real variable is the time to first crack initiation. This time is returned to the calling routine.

 $\underline{N}$  - This integer variable, passed from the calling routine, identifies the random number stream to be used.

RN - This real variable is a uniformly distributed random number.

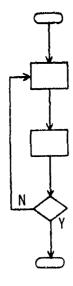
THIRD.LIFE - This real variable is the time to third crack initiation. This time is returned to the calling routine.

BETA - This real variable is the scale parameter of the fatigue life distribution.

LIFE - This real array of length three is used to temporarily hold the times to crack initiation of the three fatigue cracks.

RFL - This real variable is the element actual average fatigue life passed from the calling routine.

SECOND.LIFE - This real variable is the time to second crack initiation. This time is returned to the calling routine.



Routine FATIGUE A TEE CATTER.

Draw uniformly distributed random number.

Calculate time to crack initiation.

Times calculated for the racks?

Return.

### 8. INSTALL. MODIFICATION

### 8.1 Description

This routine represents the installation of a structural modification caused by a fatigue test failure or by aircraft service experience. The modification is installed during a repair or a D-level inspection. All previously scheduled defect initiations are cancelled, and new times to defect initiations are calculated for each aircraft when it is modified. This routine can be called from the events REPAIR and D.LEVEL.INSPECTION.

### 8.2 Local Variables

<u>DEFECT.LIFE</u> - This real variable is the time to crack initiation drawn from a distribution of fatigue lives of elements having production defects.

HOURS.TO.CORROSION - This real variable contains the value returned by routine PREDICT.CORROSION.

RST - This real variable is the remaining service time to retirement of the aircraft being considered.

STD.FAST - This real variable is the standard deviation of the distribution of fast crack growth rates.

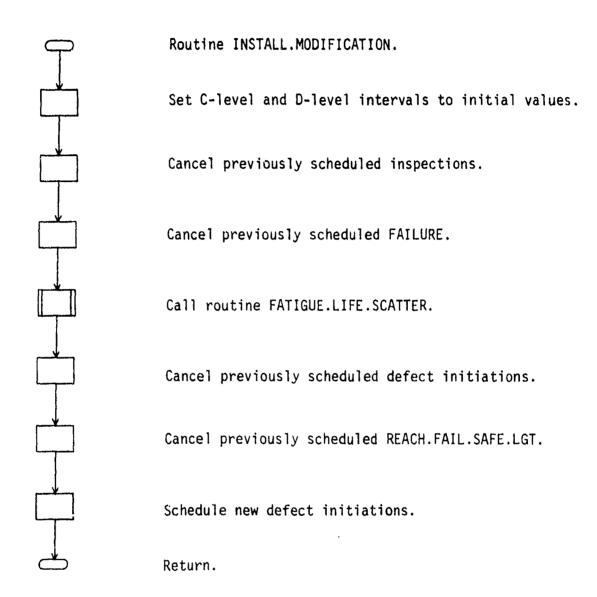
THIRD.LIFE - This real variable is the time to third crack initiation returned by routine FATIGUE.LIFE.SCATTER.

FIRST.LIFE - This real variable is the time to first crack initiation returned by routine FATIGUE.LIFE.SCATTER.

RN - This real variable is drawn from a uniform distribution of random numbers between 0 and 1.

SECOND.LIFE - This real variable is the time to second crack initiation returned by routine FATIGUE.LIFE.SCATTER.

STD.SLOW - This real variable is the standard deviation of the distribution of slow crack growth rates.



# 9. IN.SERVICE.DAMAGE

### 9.1 Description

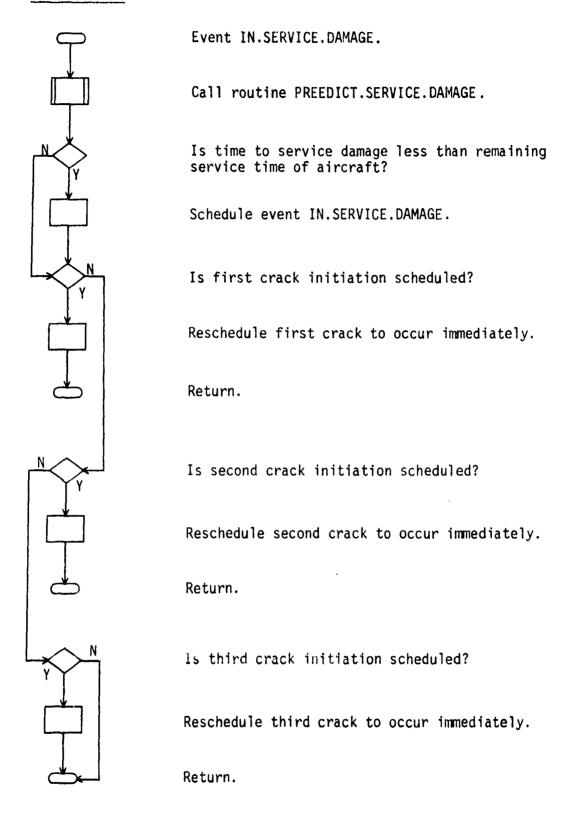
This event represents the occurrence of a service damage defect. This occurrence results in the immediate initiation of the next scheduled crack. A new time to service damage is determined. If the new time is less than the remaining time in service of the aircraft, this event is scheduled once again. This event can be scheduled from within itself or in event ENTER.SERVICE.

### 9.2 Local Variables

IDSDM - This integer variable is the identification number of the aircraft for which the event was scheduled.

RST - This real variable is the remaining service time to retirement of the aircraft being considered.

OURS.TO.SERVICE.DAMAGE - This real variable is the value returned by routine PREEDICT.SERVICE.DAMAGE.



### 10. T. IMPLEMENT. MOD

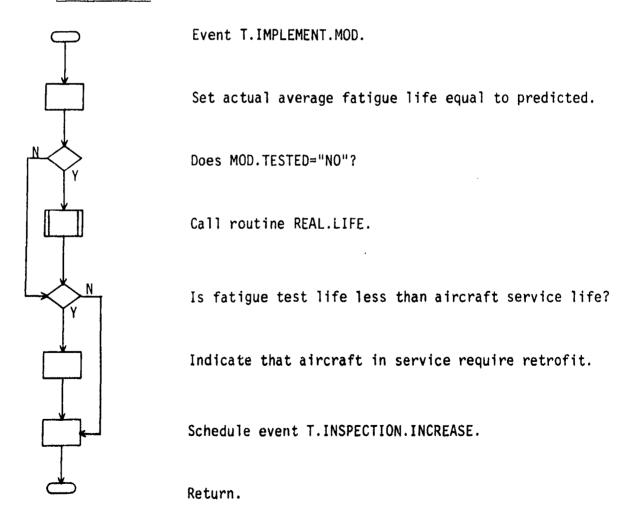
### 10.1 Description

This event represents the development of a structural modification because of a fatigue test failure. This event determines the actual average fatigue life of the modification and schedules an increase in inspection frequencies at some percentage of the fatigue test life. This event is scheduled in the MAIN program.

### 10.2 Local Variables

NSIG - This real variable is (SIG.R)(.85) and is based on the assumption that a modification usually improves the actual average fatigue life of a particular design.

NMU - This real variable is MU.R+0.15(1.0-MU.R) and is also based on the foregoing assumption for NSIG.



### 11. PREEDICT.SERVICE.DAMAGE

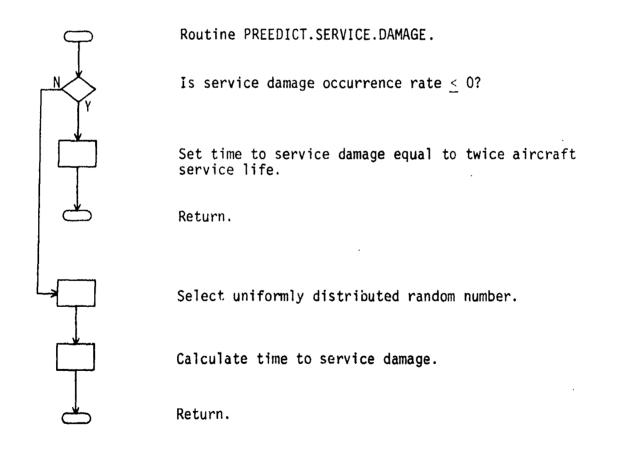
### 11.1 Description

This routine generates the time to service damage occurrence for a given aircraft from a constant service damage occurrence rate. If the service damage occurrence rate is zero in the input, the routine sets the time to service damage occurrence as twice that of the aircraft service life. This routine can be called from events ENTER. SERVICE and IN. SERVICE. DAMAGE.

#### 11.2 Local Variables

OURS.TO.SERVICE.DAMAGE - This real variable is the time to service damage occurrence in flight hours. This time is returned by the routine.

 $\overline{\text{RN}}$  - This real variable is a uniformly distributed random number between 0 and 1.



### 12. PREDICT. CORROSION

### 12.1 Description

This routine generates time to corrosion initiation for a given aircraft from a time-dependent occurrence rate approximated by two constant rates. The first constant occurrence rate, the second constant occurrence rate, and the service time on the aircraft when the second rate goes into effect are all input variables. This routine can be called from the routine INSTALL. MODIFICATION and events ENTER.SERVICE and REPAIR.

### 12.2 Local Variables

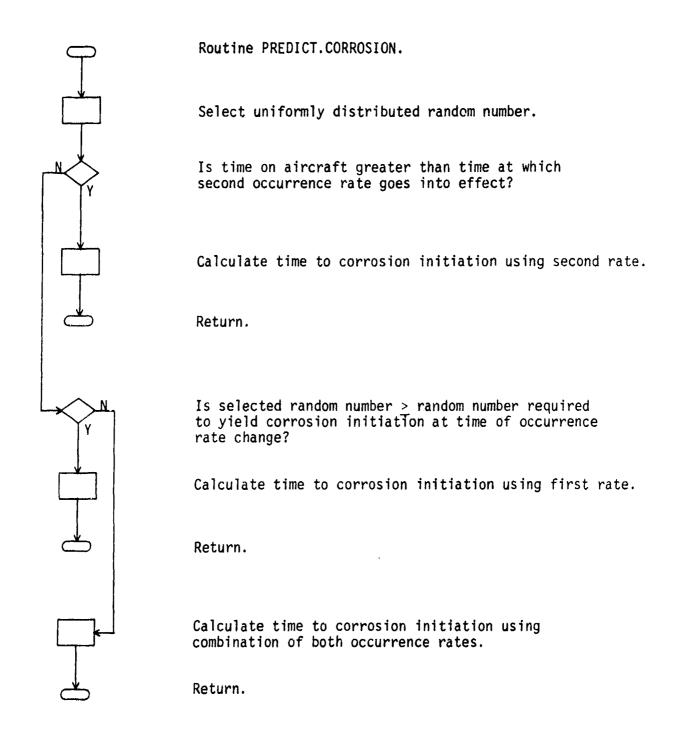
,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这一个时间,这

CRCT - This real variable is the remaining time in flight hours until the second corrosion occurrence rate goes into effect. This variable can be negative indicating that the second rate is already in effect.

 $\overline{\text{LD}}$  - This real variable is used to hold an intermediate value during the calculation of time to corrosion initiation. The calculation uses a combination of both corrosion occurrence rates.

HOURS. TO. CORROSION - This real variable is the flight time until corrosion initiation. This time is returned to the calling routine.

RN - This real available is a uniformly distributed random number.



#### 13. COROSION

### 13.1 Description

This event represents the initiation of a corrosion defect. The remaining time to crack initiation of all scheduled cracks is reduced by a corrosion damage factor. If either of the events FAILURE or REACH.FAIL.SAFE.LGT is scheduled, its remaining time until occurrence is also reduced by the corrosion damage factor. This event can be scheduled in the routine INSTALL. MODIFICATION and in events ENTER.SERVICE and REPAIR.

#### 13.2 Local Variables

CDM. MULTIPLYING. FACTOR - This real variable is the factor which when multiplied by the remaining time to crack initiation accounts for the shortening effect of corrosion on fatigue lives.

NFTM - If a FAILURE has been scheduled, this real variable is the remaining time until its occurrence.

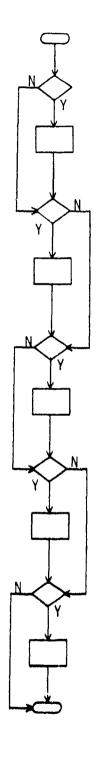
REMAINING.LIFE - This real variable is the remaining time until a scheduled crack initiation.

TRT - If a REACH.FAIL.SAFE.LGT has been scheduled, this real variable is the remaining time until its occurrence multiplied by the corrosion damage factor.

IDCO - This integer variable contains the identification number of the aircraft for which the event CORROSION was scheduled.

REDUCED. REMAINING. LIFE - This real variable is the REMAINING. LIFE multiplied by the corrosion damage factor.

RST - This real variable is the remaining service time of the aircraft under consideration.



Event COROSION.

Is event REACH.FAIL.SAFE.LGT scheduled?

Reduce remaining time to occurrence of REACH.FAIL.SAFE.LGT by corrosion damage factor.

Is event FAILURE scheduled?

Reduce remaining time to occurrence of FAILURE by corrosion damage factor.

Is first crack initiation scheduled?

Reduce remaining time to first crack initiation by corrosion damage factor.

Is second crack initiation scheduled?

Reduce remaining time to second crack initiation by corrosion damage factor.

Is third crack initiation scheduled?

Reduce remaining time to third crack initiation by corrosion damage factor.

Return.

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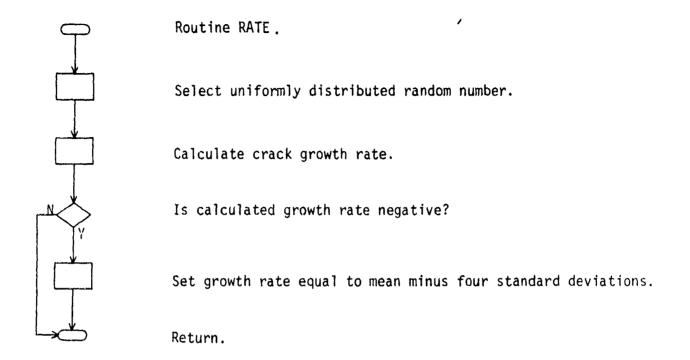
### 14. RATE

# 14.1 Description

This routine statistically generates element crack growth rates which reflect variation in material properties and load environment. The growth rates are randomly drawn from a normal distribution which is defined by a mean growth rate and a standard deviation passed from the calling routine. If a random draw yields a negative growth rate, the rate is set equal to the mean growth rate minus four standard deviations. Thus, the user must be sure that the standard deviation is always less than one-fourth of the mean. This routine is defined as a function in the routine PREAMBLE and is used in event ENTER. SERVICE and routine INSTALL. MODIFICATION.

### 14.2 Local Variables

- $\frac{G1}{In}$  This real variable is used to hold intermediate values  $\frac{G1}{In}$  the calculation of the crack growth rate.
- ${\tt M}$  This real variable is the mean crack growth rate passed from the calling routine.
- RN This real variable is a uniformly distributed random number between 0 and 1.
- S This real variable is the crack growth rate standard deviation passed from the calling routine.
- 2 This real variable is the element crack growth rate returned to the calling routine.
- G2 This real variable is used to hold intermediate values In the calculation of the crack growth rate.
- N This integer variable, passed from the calling routine, Identifies the random number stream to be used.
- RNI This real variable, equal to 1.0 RN, is a uniformly distributed random number between 0 and 1.
- W This real v viable is used to hold intermediate values In the calculation of the crack growth rate.



### 15. 1.STRENGTH.REDUCTION

### 15.1 Description

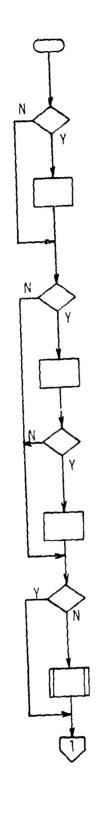
This event represents the initiation of the first crack. If the long list option is in effect, the aircraft identification number and flight hours are printed for those aircraft being tracked. A uniform random number is compared with the probability of internal cracking to determine whether this crack initiates internally. If it does initiate internally, the time until it becomes external is calculated and the event 1.ITE is scheduled. Next, the time to structural failure is calculated by using the three-part residual strength curve described in Vol. II. If this time is less than the remaining service life of the aircraft, the event FAILURE is scheduled. The time until the residual strength of the element reaches the fail-safe strength is calculated, and event REACH.FAIL.SAFE.LGT is scheduled. This event can be scheduled in events ENTER.SERVICE, INSTALL.MODIFICATION, and REPAIR.

### 15.2 Local Variables

- ARG This real variable is used as an intermediate value in the calculation of time until structural failure.
- GR2 This real variable is the fast crack growth rate.
- IDISR This integer variable is the aircraft identification number.
- K1, K13, K4, K9, LG These real variables are used as intermediate values in the calculation of time until structural failure.
- $\frac{\text{LIST}}{\text{list}}$  This real variable is set equal to 1.0 when the long  $\frac{\text{List}}{\text{list}}$  option is in effect and the aircraft being processed is one of those being tracked.
- R2 This real variable is the strength degradation rate when the crack length is between the critical crack length and the fail-safe length.
- <u>SF</u> This real variable is the element fail-safe strength.
- <u>S1</u> This real variable is the element residual strength when the crack is at the critical crack length.
- TAR This real variable is the simulation time at which the aircraft being processed retires from service.
- $\overline{11}$  This real variable is the time in flight hours for the crack to grow from its initiation to the critical crack length.

- GR1 This real variable is the element slow crack growth rate.
- K10, K12, K2, K8, LGK5 These real variables are used as intermediate values in the calculation of time until structural failure.
- R1 This real variable is the strength degradation rate from crack initiation to critical crack length.
- $\overline{R3}$  This real variable is the strength degradation rate from fail-safe strength to structural failure.
- <u>SU</u> This real variable is the ultimate strength of the element.
- $\underline{T}$  This real variable is the time in flight hours until a crack initiated internally becomes external.
- TTF This real variable is the time in flight hours until element failure.
- $\frac{T2}{the}$  This real variable is the time in flight hours until the fail-safe strength is reached.

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Event 1.STRENGTH.REDUCTION.

Is long list option in effect?

Print ID number and flight hours for aircraft being tracked.

Does crack initiate internally?

Set 1.INT to "YES".

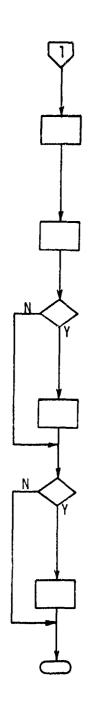
Is time until crack becomes external less than remaining service life of aircraft?

Schedule event 1.ITE.

Is aircraft presently inspected below overhaul level?

Call routine INSPECTION.SCHEDULER.

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Calculate time until residual strength equals fail-safe strength.

Calculate time until structural failure.

Is time until fail-safe strength less than remaining service life of aircraft?

Schedule event REACH.FAIL.SAFE.LGT.

Is time until failure less than remaining service life of aircraft?

Schedule event FAILURE.

#### 16. 2.STRENGTH.REDUCTION

### 16.1 Description

This event represents the second crack initiation. long list option is in effect, the aircraft identification number and flight hours are printed for those aircraft being tracked. A uniform random number is compared with the probability of internal cracking to determine whether this crack initiates internally. If it does initiate internally, the time until it becomes external is calculated and the event 2.ITE is scheduled. Next, the time to structural failure is calculated by using the same three-part residual strength curve as in event 1.STRENGTH.REDUCTION. However, the crack growth rate is now one calculated by a leastsquares fit of points determined from the sum of crack lengths of the two cracks. This calculation is described in Vol. II. If the time until failure is less than the remaining service life of the aircraft, the event FAILURE is scheduled. The time until the residual strength of the element reaches the fail-safe strength is calculated and event REACH. FAIL. SAFE. LGT is scheduled. This event can be scheduled in events ENTER.SERVICE, INSTALL.MODIFI-CATION, and REPAIR.

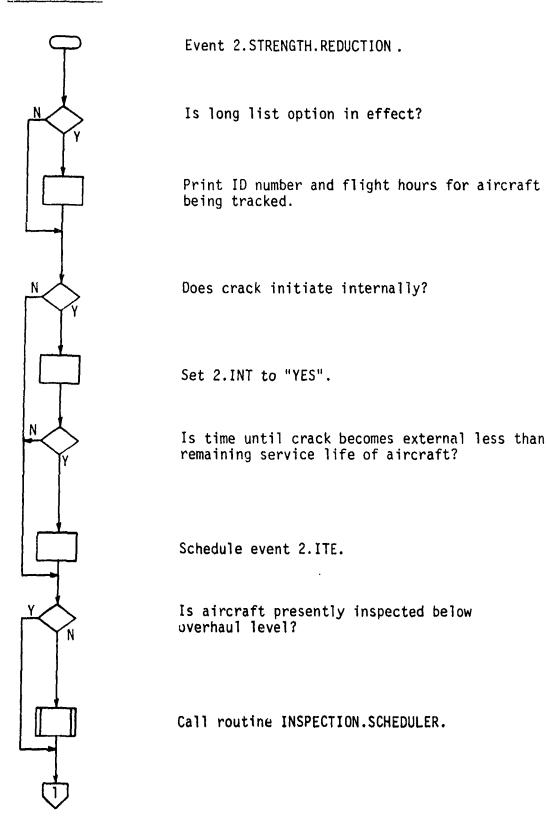
# 16.2 Local Variables

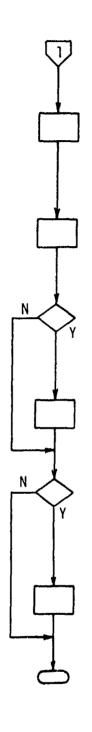
- ARG This real variable is used as an intermediate value in the calculation of time until element failure.
- CL This real variable is the crack length when the crack initiated.
- GR1 This real variable is the slow crack growth rate.
- K10, K12, K2, K8, LGK5, N These real variables are used as intermediate values in the calculation of time until element failure.
- $\overline{\text{N}2}$  This real variable is the strength degradation rate  $\overline{\text{W}h}$ en the composite crack length is between the critical crack length and the fail-safe length.
- SF This real variable is the fail-safe strength of the element.
- SU This real variable is the ultimate strength of the element.
- T This real variable is the time in flight hours until a crack initiated internally becomes external.

- TAR This real variable is the simulation time at which the aircraft being processed retires from service.
- TCL This real variable is the time it takes a single crack to grow from its initiation to the critical crack length.
- T1 This real variable is the time until the first crack reaches its critical crack length.
- W, WXS, WY, Y These real variables are used as intermediate values in the calculation of time until structural failure.
- CCL This real variable is the critical crack length of the element.
- DL This real variable is the length of the first crack at the time of corrosion initiation.
- GR2 This real variable is the fast crack growth rate of the element.
- ID2SR This integer variable is the aircraft identification number.
- K1, K11, K13, K4, K9, LG These real variables are used as intermediate values in the calculation of time until structural failure.
- $\frac{\text{LIST}}{\text{list}}$  This real variable is set equal to 1.0 when the long  $\frac{\text{list}}{\text{list}}$  option is in effect and the aircraft being processed is one of those being tracked.
- $\frac{Rl}{between}$  the time that a first crack initiates until the time that this crack reaches its critical length.
- R3 This real variable is the strength degradation rate from fail-safe strength until structural failure.
- SMW This real variable is used as an intermediate value in the calculation of time until structural failure.
- S1 This real variable is the element residual strength when the first crack is at the critical crack length.
- TAC This real variable is the time of corrosion initiation.

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- TA1 This real variable is the time of first crack initiation.
- $\underline{\text{TTF}}$  This real variable is the time in flight hours until structural failure.
- $\overline{12}$  This real variable is the time in flight hours from when an element has a residual strength until when it has a fail-safe strength.
- WX, WXY, X These real variables are used as intermediate values in the calculation of time until structural failure.





Calculate time until residual strength equals fail-safe strength.

Calculate time until structural failure.

Is time until fail-safe strength less than remaining service life of aircraft?

Schedule event REACH.FAIL.SAFE.LGT.

Is time until failure less than remaining service life of aircraft?

Schedule event FAILURE.

#### 17. 3.STRENGTH.REDUCTION

### 17.1 Description

This event represents the third crack initiation. If the long list option is in effect, the aircraft identification number and flight hours are printed for those aircraft being tracked. uniform random number is compared with the probability of internal cracking to determine whether this crack initiates internally. If it does initiate internally, the time until it becomes external is calculated and the event 3.ITE is scheduled. the time to structural failure is calculated by using the same three-part residual strength curve as in event 1.STRENGTH.REDUC-TION. However, the crack growth rate is now calculated by a least-squares fit of points determined from the sum of crack lengths of the three cracks. This calculation is described in Vol. II. If the time until failure is less than the remaining service life of the aircraft, the event FAILURE is scheduled. The time until the residual strength of the element reaches the fail-safe strength is calculated and event REACH.FAIL.SAFE.LGT is scheduled. This event can be scheduled in events ENTER. SERVICE, INSTALL. MODIFICATION, and REPAIR.

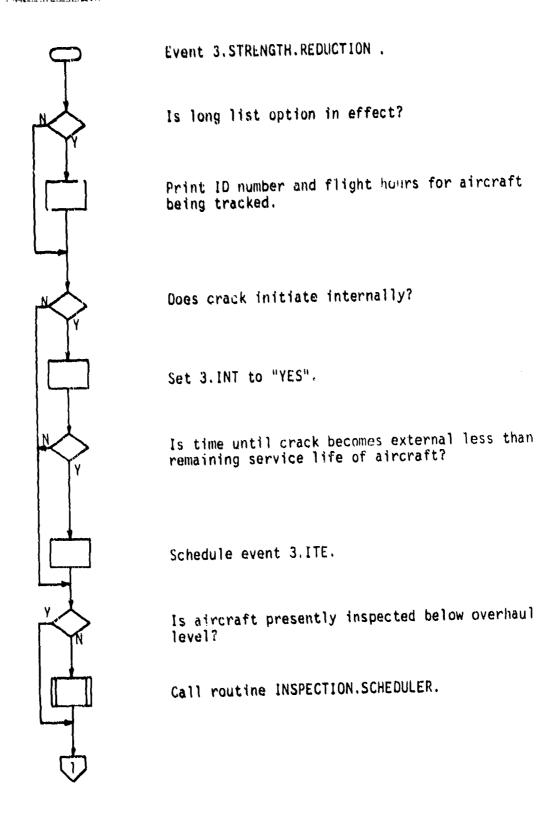
### 17.2 Local Variables

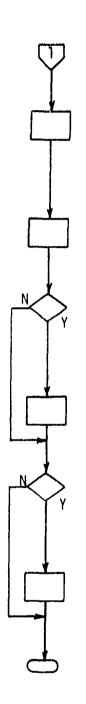
- ARG This real variable is used as an intermediate value in the calculation of time until structural failure.
- <u>CL</u> This real variable is the crack length when the crack initiated.
- GR1 This real variable is the slow crack growth rate.
- $\underline{\text{K10, K12, K2, K8, LGK5, N}}$  These real variables are used as intermediate values in the calculation of time until structural failure.
- $\frac{R2}{when}$  This real variable is the strength degradation rate when the composite crack length is between the critical crack length and the fail-safe length.
- SF This real variable is the fail-safe strength of the element.
- $\underline{SU}$  This real variable is the ultimate strength of the element.
- $\underline{T}$  This real variable is the time in flight hours until a crack initiated internally becomes external.
- TAR This real variable is the simulation time at which the aircraft being processed retires from service.

- TA2 This real variable is the time of the second crack initiation.
- TTF This real variable is the time in flight hours until structural failure.
- $\frac{T2}{when}$  This real variable is the time in flight hours from when the element has a residual strength until it reaches a fail-safe strength.

- $\underline{WX}$ ,  $\underline{WXY}$ ,  $\underline{X}$ ,  $\underline{Y}$  These real variables are used as intermediate values in the calculation of time until structural failure.
- <u>1CL</u> This real variable is the crack length of the first crack at third crack initiation.
- <u>CCL</u> This real varaible is the critical crack length of the element.
- DL This real variable is the length of the first crack at the time of corrosion initiation.
- GR2 This real variable is the fast crack growth rate of the element.
- ID3SR This integer variable is the aircraft identification number.
- K1, K11, K13, K4, K9, LG These real variables are used as intermediate values in the calculation of time until structural failure.
- LIST This real variable is set equal to 1.0 when the long list option is in effect and the aircraft being processed is one of those being tracked.
- Rl This real variable is the strength degradation rate from when the first crack initiates until it reaches its critical crack length.
- R3 This real variable is the strength degradation rate from fail-safe strength until structural failure.
- SMW This real variable is used as an intermediate value in the calculation of time until structural failure.
- S1 This real variable is the element residual strength when the first crack is at the critical crack length.

- TAC This real variable is the time of corrosion initiation.
- $\frac{TAl}{tion}$ . This real variable is the time of first crack initiation.
- TCL This real variable is the time it takes a single crack to grow from its initiation to the critical crack length.
- $\underline{T1}$  This real variable is the time until the first crack reaches its critical crack length.
- W, WXS, WY, XK, Y2 These real variables are used as intermediate values in the calculation of time until structural failure.
- <u>2CL</u> This real variable is the length of the second crack at the time of the third crack initiation.





Calculate time until residual strength equals fail-safe strength.

Calculate time until structural failure.

Is time until fail-safe strength less than remaining service life of aircraft?

Schedule event REACH.FAIL.SAFE.LGT.

Is time until failure less than remaining service life of aircraft?

Schedule event FAILURE.

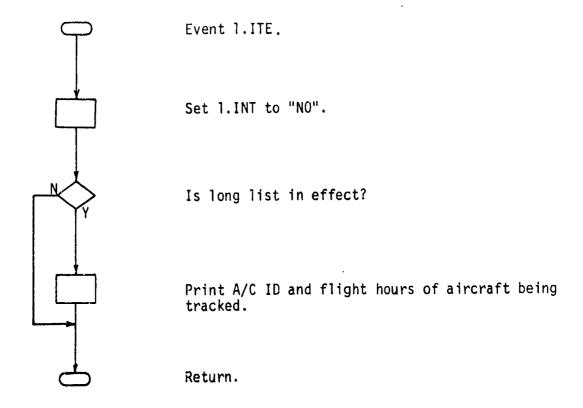
#### 18. 1.ITE

# 18.1 Description

This event represents the time when a first crack which initiated internally becomes external. This time is defined as the time when the element crack length reaches a percentage of the critical crack length. This percentage is an input parameter. At this time the appropriate element of the alpha array 1. INT is changed from "YES" to "NO". If the long list option is in effect, the aircraft identification and flight hours are printed for those aircraft being tracked. This event is scheduled in event 1. STRENGTH. REDUCTION.

# 18.2 Local Variables

IDIE - This integer variable is the aircraft identification number.



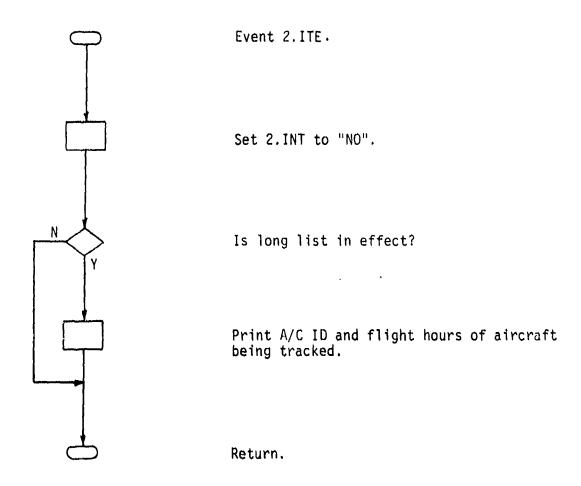
# 19. 2.ITE

### 19.1 Description

This event represents the time when a second crack which initiated internally becomes external. This time is defined as the time when the element crack length reaches a percentage of the critical crack length. This percentage is an input parameter. At this time the appropriate element of the alpha array 2.INT is changed from "YES" to "NO". If the long list option is in effect, the aircraft identification and flight hours are printed for those aircraft being tracked. This event is scheduled in event 2.STRENGTH.REDUCTION.

### 19.2 Local Variables

IDZE - This integer variable is the aircraft identification number.



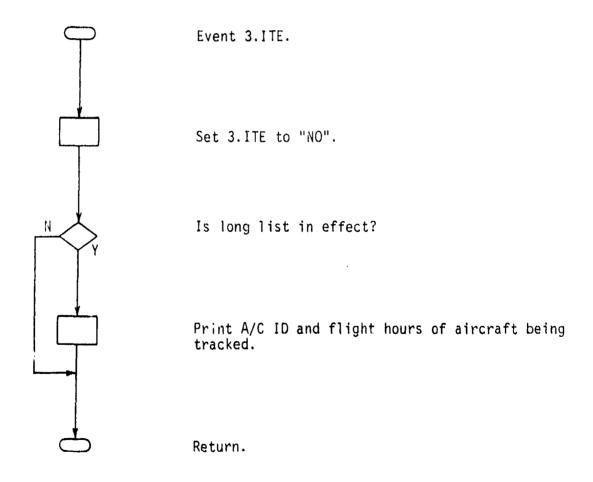
### 20. 3.ITE

### 20.1 Description

This event represents the time when a first crack which initiated internally becomes external. This time is defined as the time when the element crack length reaches a percentage of the critical crack length. This percentage is an input parameter. At this time the appropriate element of the alpha array 3.INT is changed from "YES" to "NO". If the long list option is in effect, the aircraft identification and flight hours are printed for those aircraft being tracked. This event is scheduled in event 3.STRENGTH.REDUCTION.

### 20.2 Local Variables

ID3E - This integer variable is the aircraft identification number.



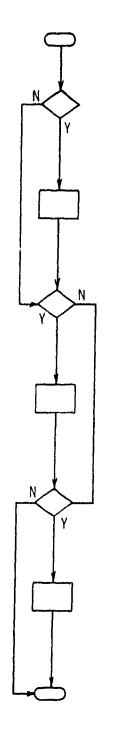
### 21. INSPECTION. SCHEDULER

# 21.1 Description

This routine schedules all inspections below the overhaul level on a given aircraft. To conserve execution time, the inspections are scheduled so that the aircraft is not inspected before the defect reaches its minimum detectable size at each level of inspection. This routine is called from events 1. STRENGTH. REDUCTION and COROSION.

### 21.2 Local Variables

- C1 This real variable is the corrosion growth rate used to calculate the time to the minimum detectable corrosion area.
- $\overline{N}$  This integer variable indicates whether a crack initiation or a corrosion initiation caused this routine to be called.
- TML This real variable is the time to the minimum detectable defect size calculated for each level of inspection.
- $\frac{M1}{ca}$  This real variable is the crack growth rate used to calculate the time to the minimum detectable crack length.
- S.INSP.AT This real variable is the simulation time at which the first inspection at each level is scheduled.



Routine INSPECTION. SCHEDULER.

Is lowest external inspection level less than or equal to A-level?

Calculate time to minimum detectable size. Schedule A-level inspection.

Is lowest external inspection level less than or equal to B-level?

Calculate time to minimum detectable size. Schedule B-level inspection.

Is lowest external inspection level less than or equal to C-level?

Calculate time to minimum detectable size. Schedule C-level inspection.

### 22. A.LEVEL.INSPECTION

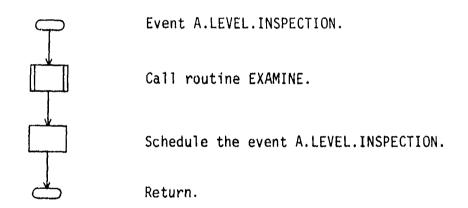
### 22.1 Description

This event represents the performance of an A-level inspection. The constants which define the probability of detection equation at the A-level are passed to the routine EXAMINE which handles the actual inspection calculations for all levels of inspection. This event can be scheduled in the routine INSPECTION. SCHEDULER and the events A.LEVEL.INSPECTION, B.LEVEL.INSPECTION, C.LEVEL.INSPECTION, and D.LEVEL.INSPECTION.

# 22.2 Local Variables

FOUND - Not used in this event, this real variable is returned by routine EXAMINE with a non-zero value whenever a defect is found.

IDA - This integer variable is the identification number of the aircraft being inspected.



# 23. B.LEVEL.INSPECTION

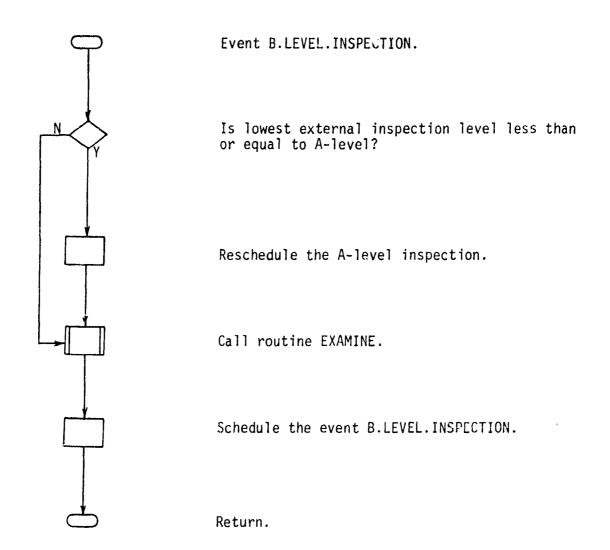
# 23.1 Description

This event represents the performance of a B-level inspection. The constants which define the probability of detection equation at the B-level are passed to the routine EXAMINE. The inspection interval for each inspection level is not necessarily an even multiple of all lower level intervals. If an A-level inspection is scheduled, it is cancelled and rescheduled at present time plus one A-level interval later. This event can be scheduled in the routine INSPECTION. SCHEDULER and the events B.LEVEL.INSPECTION, C.LEVEL.INSPECTION.

### 23.2 Local Variables

FOUND - Not used in this event, this real variable is returned by routine EXAMINE with a non-zero value whenever a defect is found.

IDB - This integer variable is the identification number of the aircraft being inspected.



### 24. C.LEVEL.INSPECTION

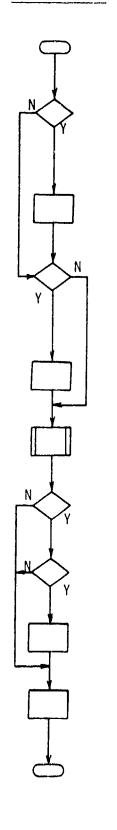
### 24.1 Description

This event represents the performance of a C-level inspection. If there is either an A-level or a B-level inspection currently scheduled, it is cancelled and rescheduled at one A-level interval or B-level interval, respectively, later. The constants which define the probability of defect detection equation at the C-level are passed to the routine EXAMINE. If a crack is detected and the aircraft is one of the ten high-time aircraft, the time of detection is stored. This event can be scheduled in the routine INSPECTION. SCHEDULER and the events C.LEVEL. INSPECTION and D.LEVEL. INSPECTION.

### 24.2 Local Variables

FOUND - This real variable is returned by routine EXAMINE with a value of one whenever a crack is detected.

IDC - This integer variable is the identification number of the aircraft being inspected.



Event C.LEVEL.INSPECTION.

Is lowest external inspection level less than or equal to B-level?

Reschedule B-level inspection.

Is lowest external inspection level less than or equal to A-level?

Reschedule A-level inspection.

Call routine EXAMINE.

Was crack detected?

Is aircraft one of ten high-time aircraft?

Retain time of detection.

Schedule event C.LEVEL.INSPECTION.

### 25. D.LEVEL.INSPÉCTION

### 25.1 Description

This event represents the performance of a D-level inspection. If the long list option is in effect on this element, the identification number and flight hours are printed for each aircraft being tracked. If there are any lower level inspections currently scheduled, they are cancelled and rescheduled at one inspection interval later. The constants which define the probability of defect detection equation at the D-level are passed to the routine EXAMINE. The defect histories of the ten high-time aircraft are now examined. If all of the ten high-time aircraft have gone one D-level interval without any cracks detected at either the C-level or the D-level, then the C-level and D-level inspection intervals are increased by the input factor FREQ. DECREASE.

Any pending modifications are installed at this time. This event can be scheduled in the events ENTER. SERVICE and D. LEVEL. INSPECTION.

### 25.2 Local Variables

FOUND - This real variable is returned by routine EXAMINE with a value of one whenever a crack is detected.

IDD - This integer variable is the identification number of the aircraft being inspected.

Event D.LEVEL.INSPECTION.

Is long list option in effect on this element?

Is this aircraft being tracked?

Print aircraft number and flight hours.

Are lower level inspections scheduled?

Reschedule lower level inspections.

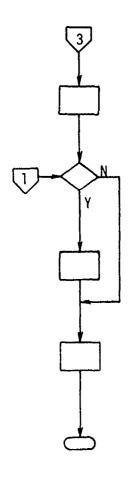
Call routine EXAMINE.

Is aircraft one of ten high-time aircraft?

Was crack detected?

Retain time of detection.

Have all ten high-time aircraft gone one D-level interval without any cracks detected at either C-level or D-level?



Increase C-level and D-level inspection intervals.

Is a modification pending on this aircraft?

Install modification.

Schedule event D.LEVEL.INSPECTION.

#### 26. EXAMINE

# 26.1 Description

This routine performs the numerical comparison which determines whether a defect is detected at each level of inspection. The constants which define the probability of defect detection equation are passed to this routine from the event: which represent the different levels of inspection. If the long 1: toption is in effect, each time a defect is detected on one of the aircraft being tracked, the following are printed: the size of the defect, the inspection level, the aircraft identification number, and the number of flight hours on the aircraft. This routine can be called from events A.LEVEL.INSPECTION, B.LEVEL.INSPECTION, C.LEVEL.INSPECTION, and D.LEVEL.INSPECTION.

### 26.2 Local Variables

- AREA This real variable is the calculated area of the corrosion defect.
- <u>CL</u> This real variable is the calculated length of the fatigue cracks.
- LIST This real variable is set equal to 1.0 if the long list option is in effect and the aircraft being inspected is one of those being tracked.
- $\frac{M2}{the}$  This real variable is the fast crack growth rate for the aircraft being inspected.
- TAC This real variable is the simulation time of the corrosion initiation.
- TA2 This real variable is the simulation time of the second crack initiation.
- $\overline{XA}$  This real variable is one of the probability of detection equation constants passed by the calling event.
- YA This real variable is one of the probability of detection equation constants passed by the calling event.
- $\frac{Z}{P}$  This alpha variable is the level of inspection being performed.
- ZL This real variable is one of the probability of detection equation constants passed by the calling event.
- <u>CCL</u> This real variable is the critical crack length of the element.

- FOUND This real variable is set equal to two whenever corrosion is detected and to one whenever a crack is detected.
- M1 This real vairable is the slow crack growth rate for the aircraft being inspected.
- $\underline{N}$  This integer variable identifies the inspection level and is passed by the calling event.
- TA1 This real variable is the simulation time of the first crack initiation.
- $\frac{TA3}{c\,rack}$  This real variable is the simulation time of the third  $\frac{c\,rack}{c\,rack}$  initiation.
- XL This real variable is one of the probability of detection equation constants passed by the calling event.
- YL This real variable is one of the probability of detection equation constants passed by the calling event.
- $\overline{ZA}$  This real variable is one of the probability of detection equation constants passed by the calling event.

Routine EXAMINE.

Does corrosion exist?

Is the area inspected?

Is random number less than probability of detection?

Set tally counters to defect size. Let FOUND=2.

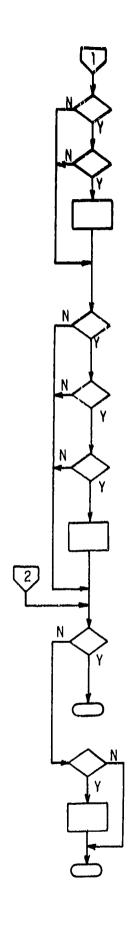
Does first crack exist?

Is the area inspected?

Is random number less than probability of detection?

Set tally counters to defect size. Let FOUND=1.

Does second crack exist?



Is the area inspected?

Is random number less than probability of detection?

Set tally counters to defect size. Let FOUND=1.

Does third crack exist?

Is the area inspected?

Is random number less than probability of detection?

Set tally counters to defect size. Let FOUND=1.

Was routine called from D.LEVEL.INSPECTION?

Return.

Is FOUND greater than zero?

Schedule event REPAIR.

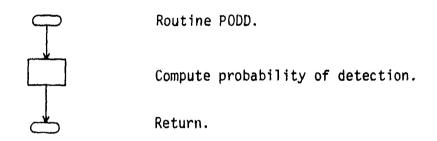
### 27. PODD

### 27.1 Description

This routine, defined as a function in the PREAMBLE, computes the probability of detecting a crack or corrosion defect of a given size. This probability is returned to the calling routine. PODD is called from routine EXAMINE and event IMMEDIATE. FLEET.INSPECTION.

# 27.2 Local Variables

- $\underline{L}$  This real variable is the size of the defect under consideration.
- $\underline{\underline{Y}}$  This real variable is an empirically determined equation constant for each level of inspection.
- $\frac{X}{t}$  This real variable is the maximum probability of detection at a given inspection level.
- $\underline{z}$  This real variable is the minimum defect size detectable at a given inspection level.



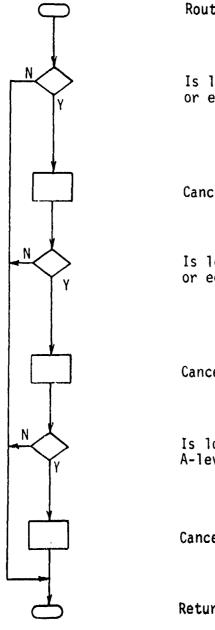
# 28. CANCEL.SCHEDULED.INSPECTIONS

# 28.1 Description

This routine cancels all scheduled inspections below the overhaul level on a given aircraft. Whenever a defect is detected and repaired, it is assumed that all other defects existing on that particular element are also repaired. This routine is called at this time to cancel all subsequent inspections. Also, if an element fails or an aircraft with existing defects is retired, this routine is called to cancel all scheduled inspections. This routine can be called from events FAILURE, RETIRE. FROM. SERVICE, and REPAIR.

# 28.2 Local Variables

There are no local variables in this routine.



Routine CANCEL.SCHEDULED.INSPECTIONS.

Is lowest level external inspection less than or equal to C-level?

Cancel the C-level inspection.

Is lowest level external inspection less than or equal to B-level?

Cancel the B-level inspection.

Is lowest level external inspection equal to A-level?

Cancel the A-level inspection.

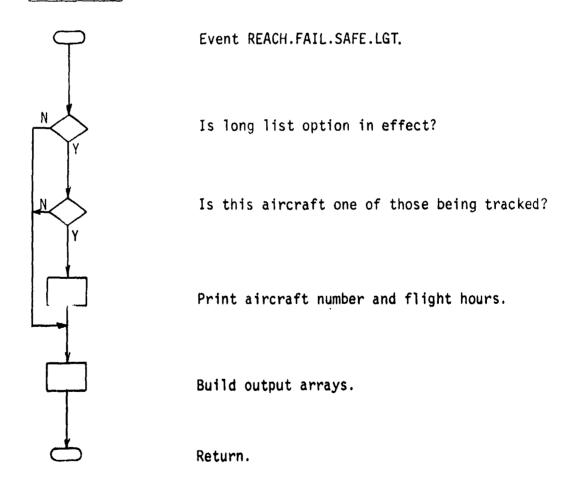
# 29. REACH.FAIL.SAFE.LGT

# 29.1 Description

This event represents the time when the residual strength of the element has been reduced to the fail-safe strength. The time and aircraft identification number are saved as part of the output. The calculation of the strength reduction is based on the sum of all crack lengths in the element. This event can be scheduled in events 1.STRENGTH.REDUCTION, 2.STRENGTH.REDUCTION, and 3.STRENGTH.REDUCTION.

# 29.2 Local Variables

IDRFS - This integer variable is the identification number of the aircraft being processed.



### 30. FAILURE

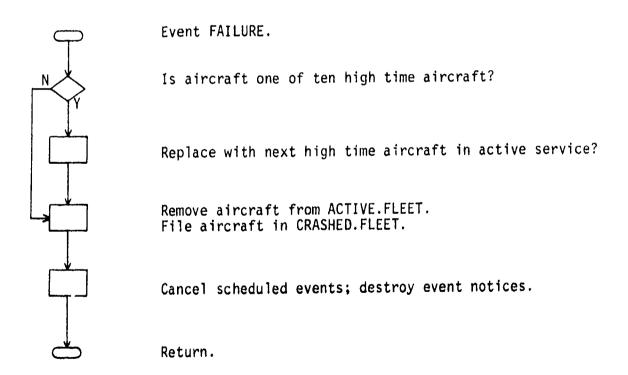
### 30.1 Description

This event represents structural failure. When this event occurs, the aircraft is removed from the active fleet. If this aircraft was one of the ten high-time aircraft being monitored for the purpose of increasing inspection intervals, it is replaced by the next high-time aircraft in active service. Any remaining scheduled events are cancelled and their event notices destroyed. This event can be scheduled in events 1.STRENGTH. REDUCTION, 2.STRENGTH.REDUCTION, and 3.STRENGTH.REDUCTION.

# 30.2 Local Variables

HOLD - This integer variable serves as an intermediate storage for aircraft identification numbers when replacing one of the ten high-time aircraft.

IDFA - This integer variable is the identification number of the aircraft under consideration.



## 31. RETIRE.FROM.SERVICE

## 31.1 Description

This event represents the retirement of an aircraft from active service. The aircraft is replaced in the HI.TIME.ACRFT array by that active aircraft not in the array which has the highest time in service. The aircraft being retired is removed from the set ACTIVE.FLEET and filed in the set FLEET.RETIRED. All remaining scheduled events for this aircraft are cancelled and the event notices destroyed. This event can only be scheduled in the event ENTER.SERVICE.

## 31.2 Local Variables

HOLD - This integer variable is used as an intermediate storage to hold aircraft identification numbers during the HI.TIME.ACRFT replacement.

IDRET - This integer variable is the identification number of the aircraft being retired from service.

#### 31.3 Flow Chart

7	Event RETIRE.FROM.SERVICE.
	Replace aircraft in HI.TIME.ACRFT array.
	Remove aircraft from ACTIVE.FLEET, file in FLEET.RETIRED.
	Cancel any scheduled events and destroy event notices.
	Return.

## 32. REPAIR

#### 32.1 Description

This event represents the structural repair of an element. If there is a modification pending on the element, it is installed at this time. If the events FAILURE and REACH.FAIL. SAFE.LGT are scheduled, they are cancelled. It is assumed that all existing defects are repaired and that new times to defect occurrences are determined in the same manner as when the aircraft entered service. The size and number of all existing cracks are compared with the inspection interval decrease criteria. If these criteria are met, then the events which decrease inspection intervals and perform special fleet-wide inspections are scheduled. All defects that were scheduled but had not occurred by this time are not affected by this event and will occur as originally scheduled. This event can be scheduled in events D.LEVEL.INSPECTION, EXAMINE, and IMMEDIATE.FLEET.INSPECTION.

#### 32.2 Local Variables

AAFL - This real variable is the actual average fatigue life of the element design.

<u>CL</u> - This real variable is the calculated fatigue crack length.

HOURS.TG.CORROSION - This real variable is the time in flight hours to corrosion initiation.

 $\underline{MAX.CRK}$  - This real variable is the maximum crack length in the element.

RST - This real variable is the remaining service time of the aircraft.

STR.RED - This real variable is the element strength reduction because of all existing cracks.

TAL - This real variable is the simulation time of the first crack initiation.

 $\frac{TA3}{crack}$  - This real variable is the simulation time of the third  $\frac{TA3}{crack}$  initiation.

<u>CCL</u> - This real variable is the element critical crack length.

FIRST.LIFE - This real variable is the time in flight hours to first crack initiation.

IDREP - This integer variable is the aircraft identification number.

POT.CRK - This real variable is the sum of the maximum crack length and the length that the crack will grow during the shortest internal inspection interval.

SECOND.LIFE - This real variable is the time in flight hours to the second crack initiation.

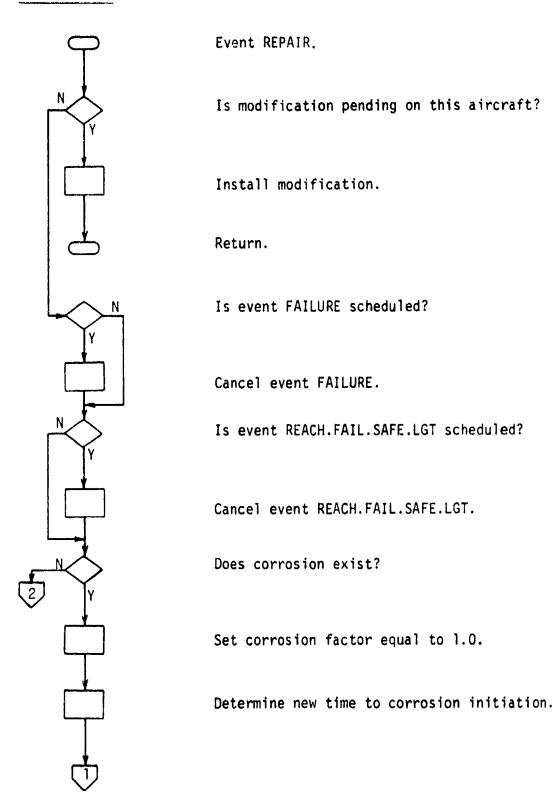
 $\underline{\text{TAC}}$  - This real variable is the simulation time of the corrosion initiation.

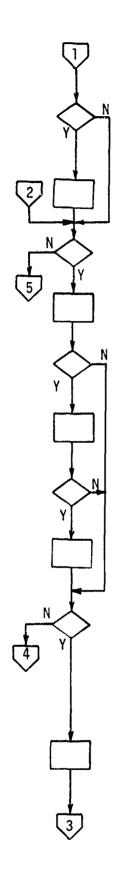
TA2 - This real variable is the simulation time of the second crack initiation.

THIRD.LIFE - This real variable is the time in flight hours to the third crack initiation.

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## 32.3 Flow Chart





Is time to corrosion initiation less than remaining service life of aircraft?

Schedule corrosion initiation.

Does first crack exist?

Calculate crack length.

Does second crack exist?

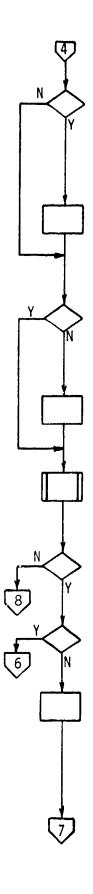
Calculate crack length.

Does third crack exist?

Calculate crack length.

Is sum of crack lengths plus crack growth rate times inspection interval greater than one-half the fail-safe crack length?

Increase inspection frequency and perform a special fleet wide inspection.



Is sum of crack lengths found in entire fleet greater than one-fifth of fleet size times fail-safe crack length?

Increase inspection frequency and perform a special fleet wide inspection.

Is there a modification pending somewhere in the fleet?

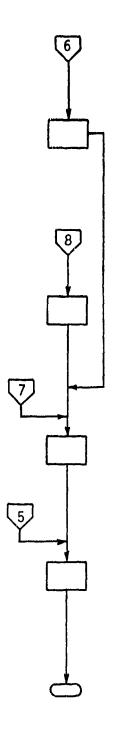
Schedule event DECISION.ON.MOD.

Call routine FATIGUE.LIFE.SCATTER.

Does second crack exist?

Does third crack exist?

Let time to first crack initiation be set to time to third crack initiation and times to second and third crack initiations be set to shortest times from routine FATIGUE.LIFE.SCATTER.



是一个时间,我们就是这个时间,我们就是这个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就会会看到这一个时间 第一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们就是一个时间,我们

Let times to all three crack initiations be taken from FATIGUE.LIFE.SCATTER.

Set time to first crack to time to second crack. Set time to second crack to time to third crack. Set time to third crack from FATIGUE.LIFE.SCATTER.

Reschedule crack initiations for those cracks with times to crack initiation less than remaining service life of aircraft.

Cancel scheduled inspections below overhaul level.

Return.

## 33. T. INSPECTION: INCREASE

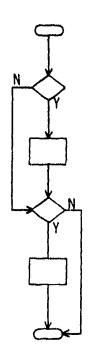
## 33.1 Description

This event represents an inspection frequency increase for a particular aircraft pending a structural modification because of a fatigue test failure. The factor by which the inspection frequency for both the close internal and the close external inspection is increased depends on the element being considered and is an input parameter. This event can only be scheduled in event T.IMPLEMENT.MOD.

## 33.2 Local Variables

IDTI - This integer variable is the identification number of the aircraft under consideration.

## 33.3 Flow Chart



Event T. INSPECTION. INCREASE.

Is lowest internal level or lowest external level of inspection equal to C-level?

Increase C-level inspection frequency.

Is lowest internal level or lowest external level of inspection equal to D-level?

Increase D-level inspection frequency.

Return.

## 34. INCREASE. INSPECTION. FREQUENCY

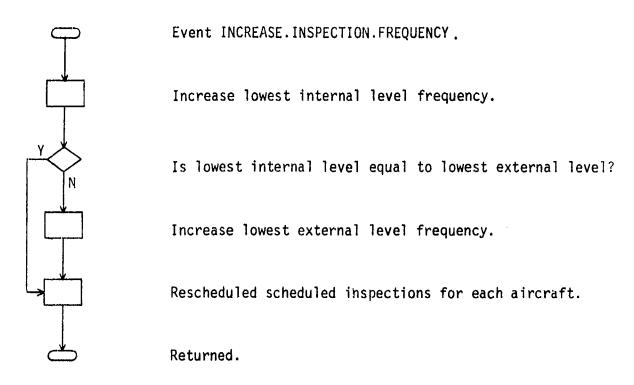
## 34.1 Description

This event represents a fleet-wide increase in the frequencies of the lowest level internal and external inspections. These may or may not be the same levels. However, the A-level and B-level frequencies are never changed. All scheduled inspections are cancelled and rescheduled to reflect the frequency increase. This event can be scheduled in events REPAIR and IMME-DIATE.FLEET.INSPECTION.

#### 34.2 Local Variables

DIFF - This real variable is the difference in flight hours between the old and new lowest internal level inspection intervals.

#### 34.3 Flow Chart



## 35. IMMEDIATE.FLEET.INSPECTION

#### 35.1 Description

This event represents an immediate fleet-wide inspection caused by finding a defect considered too hazardous to depend on scheduled inspections for detection of additional defects. Existing crack lengths and corrosion areas are calculated along with the associated probabilities of detection. As in the scheduled inspections, these probabilities are compared with a random number to determine whether or not the defect is detected. This event is always preceded by the event INCREASE. INSPECTION. FREQUENCY. Defects found during this inspection can cause an additional increase in the frequency of normally scheduled inspections. This event can only be scheduled in event REPAIR.

## 35.2 Local Variables

- AREA This real variable is the area in sq. inches of an existing corrosion defect.
- $\frac{CL}{ing}$  This real variable is the length in inches of an existing fatigue crack.
- $\underline{\text{M1}}$  This real variable is the slow crack growth rate for a particular aircraft.
- POT.CRK This real variable is the sum of the maximum crack length detected on a particular aircraft plus the product of the average crack growth rate and the current lowest internal level inspection interval.
- <u>SR</u> This real variable is the sum of the lengths of all cracks detected on a particular aircraft.
- $\underline{TAC}$  This real variable is the time of initiation of an existing corrosion defect.
- $\overline{\text{TA2}}$  This real variable is the time of initiation of an existing second fatigue crack.
- <u>CCL</u> This real variable is the critical crack length of the element under consideration.
- FOUND This real variable serves as a switch which is set equal to two when corrosion is detected and set equal to one when a crack is detected.
- MAX.CRK This real variable is the maximum crack length detected on a particular aircraft.

 $\underline{\text{M2}}$  - This real variable is the fast crack growth rate for a particular aircraft.

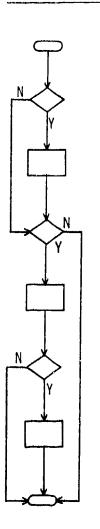
POT.STR.RED - This real variable is the sum of the detected crack lengths on a particular aircraft plus the product of the crack growth rate and the current lowest internal level inspection interval.

STR.RED - This real variable is equal to SR.

TA1 - This real variable is the time of initiation of an existing first fatigue crack.

TA3 - This real variable is the time of initiation of an existing third fatigue crack.

#### 35.3 Flow Chart



Event IMMEDIATE.FLEET.INSPECTION.

Does corrosion exist?

Calculate corrosion area and test for detection.

Do cracks exist?

Calculate crack lengths and test for detection.

Do detected cracks meet criteria for inspection frequency increase?

Schedule event INCREASE.INSPECTION.FREQUENCY to occur immediately.

Return.

## 36. DECISION.ON.MOD

## 36.1 Description

This event makes the decision on whether or not to develop a structural modification because of service experience. The decision to develop a modification is made by comparing the cost per flight hour of the modification with the repair cost per flight hour plus the increased inspection cost per flight hour. The modification cost per flight hour is found by dividing the total fleet modification cost by the remaining service life of the fleet. The repair cost per flight hour is found by dividing the total fleet repair costs since the last modification by the fleet flight time since the last modification. The increased inspection cost per flight hour is found by dividing the projected increased inspection costs by the remaining service life of the fleet. A modification is justified when

MCPH < RCPH + ICPH

where

MCPH = modification cost per flight hour

RCPH = repair cost per flight hour

ICPH = increased inspection cost per flight hour

This event can be scheduled only in event REPAIR.

## 36.2 Local Variables

ACCUMULATED. HRS - This real variable is the total fleet time since the last modification.

MD.COST - This real variable is the cost of installing a modification on a single aircraft. The costs for additional modifications can differ from those for the initial modifications.

 $\frac{NFTS}{all}$  - This real variable is the total production time of  $\frac{all}{all}$  aircraft entering service after the second production rate goes into affect.

POST. MOD. HRS - This real variable is the total fleet service time remaining after the modification.

TOOLING - This real variable is the tooling cost in the development of a modification. The costs for additional modifications can differ from those for the initial modifications.

ICPH - This real variable is the increased inspection cost per flight hour.

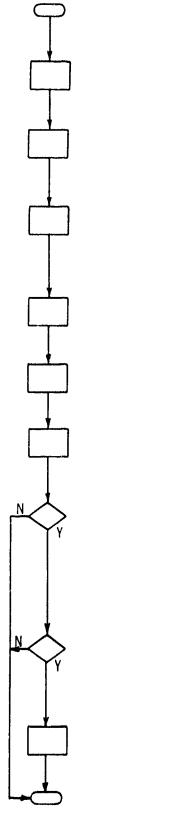
MCPH - This real variable is the modification cost per flight hour.

 $\underline{\mathsf{MRFH}}$  - This real variable is the service time remaining on a particular aircraft after its modification.

 $\underbrace{\text{NPDL}}_{\text{have}}$  - This real variable is the number of aircraft which have entered service.

 $\underline{RCPH}$  - This real variable is the repair cost per flight hour of the fleet.

## 36.3 Flow Chart



Event DECISION.ON.MOD.

Accumulate remaining service life of active fleet.

Determine fleet flight time since last modification.

Add service life of aircraft not yet produced to remaining service life of active fleet.

Calculate increased inspection costs per flight hour.

Calculate modification cost per flight hour.

Calculate repair cost per flight hour.

Is repair cost per hour plus increased inspection cost per hour greater than modification cost per hour?

Can modification be developed before last aircraft in fleet retires?

Schedule event IMPLEMENT.MODIFICATION.

Return.

#### 37. IMPLEMENT. MODIFICATION

## 37.1 Description

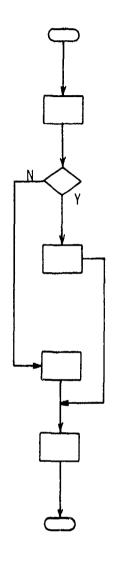
This event represents the development of a modification because of service experience. If the modification is to be fatigue tested, the actual average fatigue life is set equal to the original predicted life of the element design. Otherwise, the actual average fatigue life is determined by calling event REAL.LIFE. Elements of the alpha array SMOD.PENDING are set equal to "YES" to indicate which aircraft have service modifications pending. These modifications will be installed at the next D-level inspection or defect repair. Intervals for all levels of inspection are set to their initial values. This event can be scheduled only in event DECISION.ON.MOD.

#### 37.2 Local Variables

NSIG - This real variable is the standard deviation of the ratio distribution passed to routine REAL.LIFE.

NMU - This real variable is the mean of the ratio distribution passed to routine REAL.LIFE.

## 37.3 Flow Chart



Event IMPLEMENT.MODIFICATION.

Set inspection intervals to initial values.

Is modification tested?

Set actual average fatigue life equal to predicted average fatigue life.

Call routine REAL.LIFE.

Set alpha array SMOD.PENDING.

Return.

## 38. DISPLAY. OUTPUT

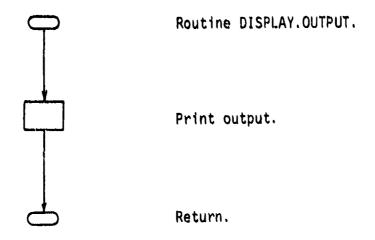
## 38.1 Description

This routine prints the standard output for each element. It is called from the MAIN program immediately after the completion of each element simulation. This output is suppressed if the long list option is in effect.

## 38.2 Local Variables

All the local variables are used to temporarily hold output values.

## 38.3 Flow Chart



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#### 39. SUMMARY

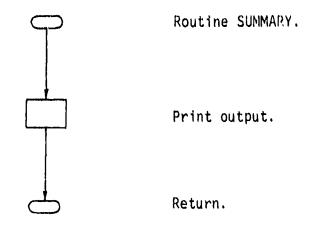
## 39.1 Description

This routine prints the standard output summary for each element type. The output of routine DISPLAY.OUTPUT for all the elements in an element type is contained in this output summary. This routine is called from the MAIN program each time a new element type is read in and at the end of the program run. This output is suppressed if the long list option is in effect.

## 39.2 Local Variables

All the local variables are used to temporarily hold output values.

## 39.3 Flow Chart



# APPENDIX B INITIAL PROGRAM SOURCE LISTING

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LINE CACT SIMSCRIPT II.5 WELEASE BF

CALL STRONG TITLE SCATTER (MFL,M) TIFLDING FIRST LIFE, SECOND. LIFE AND THIRD. LIFE	GENEMATES FLEWFUT FATIGUE LIVES RFFLFCTING RASIC FATIGUE SCATTER AND LOAD FAUTRUMMENT WARIATION	DISTRINGTON OF FATTGUE LIVES IS MEDNESERTED AS THO PARAMETER AFTRUCL ALPHA = Smade darameter Beta = Scale darameter (Characteristic value) N = Raydor vywyfr Stream		SECTATOLIFF = 1 1 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2
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AIRCHAFT
                                                    ** THIS WHITINE REPRESENTS THE INSTALLATION OF A STRUCTURAL MUDIFICATION CAUSED
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LET STO. FAST = M2. WEAN = .15

LET WSR(ID) = HAIF(M1. WEAN.SID. SLUN.5)

LET WSR(ID) = HAIF(M2. WEAN.SID. SLUN.5)

LET WSR(ID) = 141F(M2. WEAN.SID. SLUN.5)
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LET COMEP.TIMETIN = 11-4.3
LET CO.EXISTS(10) = "45"
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LET CO.EXISTS(ID) = "4N"
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CACI SIMSCRIPT II.5 RELEASE AF
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LET COMUSION = AC(TD)
IF COLEXISTS(ID) = "AS"
CATCL THE COMUSION
                          ROUTINE INSTALL, MODIFICATION
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LET FAILURE = AF(IN)
CANCEL THE FAILURF
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LT 2.CH. *KISTS(ID) NF "NG"

IF 2.CH. *KISTS(ID) NF "NG"

IF 2.CH. *KISTS(ID) NF "NG"

CANCEL THE STRENGTH, *REDUCTION

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DESTROY THE 2.STRENGTH, *REDUCTION

LE 1 5.CH. *KISTS(ID) NF "NG"

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IF I.CR.FXISIS(ID) = "NS"
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CACI SIMSCRIPT II.5 RELFASE
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\*\* APPERATES OF THE CONSTANT OCCUPARING MATES

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\*\* C.Chw.OCCUPARINCE.QATE = SECOND COMPTANT OCCUPARINCE RATE

\*\* C.Chw.WATE.CHANGE = TIME UN AIRFRANE AT CHICH SECOND RATE IS USED LET MANAS, 1)\_COMPASSION = LUG.F.F(RN) / (-1.COM.UCCUMPFNCE, RATE)
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LET 4.STRFVGW-9FOLGTIN = & $$w(fq)

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CARCEL THE 1.STHEWSTM.MEDUCTION

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ALMAYS
LET F = (CMIT,CM,LGT&PCCL)/(MSP(10)\*CGWI(ID))

IF PCCL ST 1.0

LET T = CMIT,CM\*,LGT/(MSR(ID)\*(GMI(ID)) + (CMIT,CM\*,LGT\*(PCCL+1,0))/

(PFU(ID)\*CG+1f(ID)) EVENT 1.STRENGTH.RFDUCTION(10154) SAVING THE EVENT NOTICE PMENICI 1106 1: Fall Jud Fully Flust ( 45C% 151114114) ALDAYS

IF ITMENT ILT TAR OR CONEXISTS(10) = "RS"

SCHEDUL A FITTE(15) AT TIMENT + T

LET ALCID) = 1.TF

LET (EL(10) = "YFS" LET TOWKT & TIME, V = CRMED. TIME(ID)
LET GICKM = TIME, V = CRMED. TIME(ID)
LET GICW = GAILEM + 1
LET I. MATHON = "WO"
LET I. MATHON = "WO"
LET I. MATHON = "YES"
LET I. MATHON = "YES"
LET I. MATHON = "YES"
CARCA INITATES INTERNALLY PETI-E IDISA AS AN INTEGEM VARIABLE " REPRESENTS FLAST CRACA INITIALIUM 1 St = 2.75 1 SF = 1.0 1 Med(1D) = Har-New\_F(3) 1 LG = LNG\_EF(Keb4(10)) 1 G91 = 954(15) + CG-1(1.5) LFI 1.C4.EXISIS(13) = "115"

IF 14SP.SC~(13) = ".0"

CALL INSPECTION.SC~EULER[]) HF CAMPLYSS 4444666666666 -149.

CACI SIMSCHIPT II.5 PELEASF AF

9

在在著作品是由我们的一个情况,从中的人的是是是是是是是是是这种人的人们的是是是是是是是是是是是是是是是是是是是是是是是是是是是是是是是是

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04/01/16														
	į	#ORD 29	KONO 11		WUND 17		WURD ES	MUND 45	F (340 55	MORO SX	MARD 15	#(#PD 37	TONO 24	KCIRD 21
ROUTINE														Dought F.
THIS	64 J	-: -:		٨.	¥° ₹	5 X Z	2	¥,	7° 7	<b>4</b> 1	ã	¥	ŝ <b>⊷</b>	77.
COULT VARIABLES OF	14 0504	4 050x x	ر اج	200	3000	1.120	200	1040	4045	#C20	4040	AUAD	660	SE Gette
To ange	DOURLE	TYTEGE	141FGF 141FGF	JATEGE	DOUBLE	DUCKLE DUCKLE	Distribute.	DOUGHLE	CATCH E	3 lenun	D. WALE	PERUMIE	District &	DYUME
ARG	585 1	IDISH	*	~ ×	¥ 1.1	K13	2 D	. 6	1817	~ ~ 1	S. S.	Şī	4	=

LINE CACI SIMSCRIPT II.S PFLFASE BF

STATE OF STA

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2
  09/01/76 PAGE
                                                                                                                                                                                                                             DFTERNINE MESTANAL STRENGTH MEDILITION BECAUSE OF FIRST CHACA
                EVENT 2.STRENGIM. REDUCTION (1025K) SAVING THE FUFNT NOTICE
                                            OFFINE IDASR AS AN INTEGEN VARIABLE
OFFINE x, v, n as weal, 1-Dimensional annays
ofsever n(*), v(*), a(*) as 5
LET ID x 1028H
LET ID x 1028H
LET TO X 1788H
IF TAM x EVRY, TIME(AIMPLANE(ID)) + USAGE, LIFE
IF LIMI x "VES"
                                                                                                                                                                                                                                                                   IF CO.EXISTS(I.) = "VFS"
LET CHARSIA = AC(I.)
LET TAC = IME, A(CAMASIUN)
IF TAC LE TAI
LET CL = TITE, V=TAI) = MSR(ID) + CGMI(ID)
IF CL = CT
                                                                                                                                                                                L+1 2,141(10) = "MO"

IF MATERIA, E(13) L+ 1,PW(H

LE1 2,141(13) = "VFS"

IF LIST = 1,0

PAINT 1 LINE AS FOLLONS

CACK [NITATES 141E-MALLY
                               REPRESENTS SECOND CRACK INITIATION
CACI SIMSCHIPI II.5 RELEASE AF
                                                                                             FOR I = 1 TO WIRCLION)
                                                                                                                                                                          ALFAYS
                                                                                                                                                                                                                                                                                                                                                                   :::
                                                                                                                                                                                                                               こちこのころだれん
                                                                                                                                                                                                                              -/52-
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LET II = 0.0

IF CL LI CCL

LET II = CCCL-CL) / (MSR(10) \* CGRI(ID))

LET GRI = 2.n \* MSR(10) \* CGRI(ID)

IF TAM LE IME, W + TCL

LET GR2 = 2.n \* MSR(10) + MFK(ID)) \* CGRI(ID)

JUMP AMEAD

OTHERAISE

LET X(2) = CCL + X(2) \* MFK(ID) \* CGRI(ID)

LET X(3) = CCL + X(2) \* MFK(ID) \* CGRI(ID)

LET X(3) = CCL + X(2) \* MFK(ID) \* CGRI(ID)

LET X(3) = TCL-TI

LET X(3) = TCL-TI

LET X(4) = Y(2) + (2.0 \* MFK(ID) \* CGRI(ID))

LET X(5) = TCL-TI

LET X(6) = Y(1) + WFK(ID) \* CGRI(ID)

LET X(6) = Y(1) + WFK(ID) + WFK(ID)

LET X(6) = X(6) + WFK(ID) + WFK(ID)

LET X(6) = X(6) + WFK(ID) + WFK(ID)

LET X(6) = X(6) + WFK(ID)

LET X(7) = X(6) + WFK(ID)

LET X(7) = X(7) + WFK(ID)

LET X(8) = X(1) + WFK(ID)

LET X(8) LET SU = 2.75 - (1.75 - CL) / FSAF.LGT
IF CL GT FSAF.LGT
LET SU = 1.0 - (CL-FSAF.LGT)/(LGMT.Tn.FAILUML-FSAF.LGT)
HFGARDLESS \*\* PREDICT TIME TO FAILURE FROW SECOND CHACK INITIATION LET CL = (TIME,V=TA1) \* MSR(ID)

1F CL GT CCL

LET CL = CCL + ((CL+CCL)/MSP(ID))\*MFR(ID)

REGRAPLESS

HERE LET CL = CCL + ((CL-CCL)/MSR(ID))\*MFR(ID)
JUMP AMEAD LET X(1) # 0.0 LET X(1) # 0.0 LFT X(2) # TCL JUMP AMEAD OTHERNISE 

CACT SIMSCRIPT 11.5 RELFASE AF

The same of the sa

н • R2 * II) 1.0) 3) - LÚG.E.F(ANS.F(LG+KR-K9))/кд	10 = HeRS  46 = HeSF-HeRI-12  LT = 175.0  46 = +175.0  11 = 410/(AmeExp_F(Amic))  12 = 4275.4  13 = Exp_F(Hard-211 - HeR2-12) + 1.0  14 = -106.F.F(Miia(LG+Mi2-(KM*Mi3)-M9))/M10  15 = 1.0  11 = 1.0  11 = 1.0  11 = 1.0  12 = 1.0  13 = 1.0  14 = 1.0  15 = 1.0  16 = 1.0  17 = 1.0  18 = 1.0  18 = 1.0  19 = 1.0  10 = 1.0	** ** *** *** *** *** *** *** *** ***	E. BOWLALE BOWD 27 DOUGHLE BOWD 37 DOUGHLE BOWD 37 INTEGE BOWE 45 INTEGE BOWE 1 INTEGE BOWLD 20 INTEGE BOWLD 2
F.F(A) + (H * S1 + F(H*S1))/Kl )*(FH*L-L-H*H 1].	LET HOS HARS  LET ANG = HASF-HARRAPET  LET ANG = HASF-HARRAPET  LET ANG = -175.0  LET HIS = AFFP-F(HASF-)/HIS  LET HIS = EXP_F(HASF-)/HIS  LET HIS = LOG.F.F(HISP-)/HISP  FE LIST = -LOG.F.F(HISP-)/HISP  FE LIST = LOG.F.F(HISP-)/HISP  FE FENT HISP	LET FAILURE = AF(10)  CANCEL THF FAILURE (10) AT TIME.V +  JUMP AMEAD  LSE  SCHEDUE A FAILURE (10) AT TIME.V + TIP  LET AF(10) = FAILURE  LET SM(10) = "YFS"  RELEASE  RELEASE	######################################
	LEST 410 = Heed 5 LEST 410 = H	LET FAILUME = AF(19) CANCEL THE FAILUME RESCHOULE THE FAILUME FLSE SCHEDULE A FAILURE(1 LFT AF(10) = "PES" NEWE ALAAYS NELEASE K(*), Y(*), RELEASE K*	LUCAL VANTAR DOUNLE NDRO 3 DOUNLE NDRO 3 DOUNLE NDRO 3 INTEGEW NORSO 1 INTEGEW NORSO 1 INTEGEW NORSO 2 INTEGEW NORSO 1 INTEGEW NORSO 2 INTEGEW NORSO 2 INTEGER
8.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0		- / <b>55 -</b>	4 1 0 0 1 1 1 2 1 1 1 1 1 1 1 1 1 1 1 1 1

LINE CACI SIVSCRIPT II.S WELEASE RF

II.S WELFASE AF

LINE CACI SIMSCHIOT

\*\* DETERMINE WESTOUND STREWGTH REDUCTION ANCAUSE OF FINST AND SECOND CHACKS Lf 1 = (C#II\_C#n\_LGI\*PCL)/(\u2210)\*LGP)|
if PCCL 61 i.o
LF I = C#II\_CRK\_LG1/(\u2210)\*CGMI(ID)) + (CHII\_C\u2210\u2210)|
(\u2210FI I = C#II\_CRK\_LG1/(\u2210)\*CGMI(ID)) + (CHII\_C\u2210\u2210)|
(\u2210FI I I = C#II\_D)) IP ID = TLID(LOx.1)

LEI LIST = 1.0
SAIP 1 .UIP II LINF
PHIMI 1 LI'M MITH ID, TIME, V-ENTHY, TIME (AIPPLANE (ID)) AS FOLLOWS
A/C ':), ass FXPERIENCES \$PD CHACK INITIATION AI \*\*\*\*\* HOURS
IEANE
FLSE EVENT 3.STRENGTM. REDUCTION (103SR) SAVING THE EVENT NUTICE LET 10 = 1035# LET TAR = EVTRY, TIME (AIMPLANE (10)) + 115AGE, LIFE IF LIMI = "YES" FUR I = 1 13 10 JAC (LOS) OFFINE 50554 AS AN INFEGEN VANIANCE DEFINE K. V. R AS REAL. 1-DIMENSIMAL APHRYS RESERVE K(\*). Y(\*). A(\*) AS S ALMAYS

IF TIME,V + T LT TAM OM CO.FRISTS(10) = "NS"

SCMEDULE A %.ITE(10) AT TIME,V + T

LET ASE(10) = %.TT

LET LEK(10) = "YES" LET 3.C%.EXISS(10) = "YES"

LET 1.SIMPNGTW.WE(UCITON = A1SK(10)

LET C. T. CATT.CKW.LCT

LET ICL = CATT.CKW.LCT

LET ICL = ECL.(WSW(10) + COMTION)

LET ICL = TWE A(1.SIMPNGTW. + COUCITON)

LET IAP = 11 "F. A(2.SIMPNGTW. + COUCITON) .. REPRESENTS THIAN CRACK INITIATION |F CO.FXISTS(19)= "VES" | FT CORNSIUW= AC(10) | LET FAC = TIME.A(CHRUSION) | FT FAC LF FAT ALMAYS RFGAMOLESS AL GATS 1039 1 1502252885 -157-

09/01/76 PAGE

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CACI SIVSCHIPI II.S PELEASE BF

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04/01/16
                                                                                                                                                                                                                                                                                                                                                      LFF CL = 1CL + PCL
LFF Su = 2.7% + (1.7% + CL) / FSAF.LGT
LFF Su = 1.7 + (CL-FSAF.LGT)/(LGHT.TA.FSIL1 +E+FSAF.LGT)
WEGAPOLESS
                                                                                                                                                                                                                                                                                                                                                                                                         PHEDICT TIME TI FAILURE FAUM THIRD CHACK TAITIATION
                                                                                                                   tf 1CL = (flwe,v=fai) = wSW(fD) = CGM(ff)
lf 2CL = (flwe,v=fa2) = wSR(fb) = CGM(fD)
lf 1CL & CCL = (flCL+fCL)/wSW(fD)) = wFR(fD)
wFiadin[fSS]
lf 2CL = fCL = (CCL+fCL)/wSW(fD)) = wFR(fD)
wFiadin[fSS]
lf 2CL = fCL = (fCL+fCL)/wSW(fD)) = wFR(fD)
wFiadin[fSS]
lf 2CL = fCL = fCL+fCL = fCL+fCL)/wSW(fD)) = wFR(fD)
wFiadin[fSS]
lf 2CL = fCL = fCL+fCL
                                                                                                                                                                                              HETAPOLESS

IF TAT LF TAT

LFT SCL = (TIME, V=TA1) + MSH(ID) + GG+1(ID)

IF FCL GT CCL
                                                                                                                                                                                                                                                                           LET 1CL = (TIME, V-TAI) + VSK(1D)

1P 1CL GT CCL

LEI 1CL = CCL + ((1CL-LCL)/ESW(IP)) + VFK(II)
                                                                                                                                                                                                                                                                                                       LET 2CL = (flat.v=fa2) = "Se(10)

LET 2CL to CCL

LET 2CL = CCL + (f2CL+CCL)/*Se(10))***F#(10)

**FGARDLESS
                                                                                                                                                                                                                                  LET ICL = Cil + ((ICL-CCL)/#Sk(IP))*#FK(ID)
#FGAPILESS
                                                                                                                                                                                                                                                                                                                                                                                                                                   CACT SIMSCHIPT 11.5 DELEASE NE
                                                                                                                                                                                                                                                                                                                                                                                                                          LET 11 = 3.4
                                                                                                                                                                                                                                                                                                      WEGGRANLES.
                                                                                                                                                                                                                                                                          -158 -
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If Tak LE IIuk,v + TCL
LET x(2) = r?
LET y(2) = r? + (2,0 \* MSH(ID) + MFH(ID)) + CG4[[]v)
LET y(3) = rak - Time,v
LET y(3) = y(2) + (x(3)-x(2)) + (2,0 \* MFh([]v) + MSH(IG)) + CGH[[]v)
LET y = 3
Jump anean LET X(1) = 0.0 LET X(2) = 0.0 LET X(2) = 0.0 LET X(2) = 1.0 + (sFx(10) + 2.0\*ESx(10)) + (GwJ(10)) LET X(3) = 74P + 17WE,V + 71 LET X(3) = 7(2) + (x(3)-x(2)) + (2.0\*EFx(10) + wSw(10)) + (GwJ(10)) LEI ((2) = 12 LEI x(2) = 72 4 (2.0 4 vSH(ID) + VFH(ID)) \* CAPI(ID) LEI x(3) = 7CD + (ICL-12) \* (2.0 4 wFH(ID) + VSH(ID)) \* CGHI(ID) LEI x(4) = 1AP - 114E,v LEI x(4) = 7(3) + (x(4)+ICL) \* (3.0 + NFH(ID)) \* CGHI(ID) 09/21/15 LFT X(1) = 0.0 LFT X(1) = 0.0 LFT Y(1) = 0.0 LFT TZ = (CCL-2CL)/(\*SR(ID)\*ChRJ(ID)) FFTA\* LETTAF, V + T2 LFT GW2 = (0.0 \* \*SR(ID) + \*MFR(ID)) + FGPJ(ID) LFT GW1 = 642 LET 5R2 = (WFA(IO) + 2.0\*MSk(IO)) \* CGFI(IP) JUMP AMEAD OTHERAISE OTHFRAISE LET X(2) = TCL JUMP AME AL JUND SHFED JUMP AMEAN AST HAMIN BINGHINE -159-

4 10.5

CACI SINSCHIPT II.S PELFANE NE

是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一个人,也是一

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7
   FAGE
   91/10/60
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         LEI 12 = (FSIF.LGT - CL) / (H2

REGARDLESS

LET 12 = (FSIF.LGT - CL) / (H2

REGARDLESS

LET SE = 1.0

LET SE = 1.0

LET WE = ((SI-SF)+6GP) / FSAF.LGT

LET WE = (SI-SF)+6GP) / FSAF.LGT

LET WE = 642 / (LGHT.D2-FAILUME - FSAF.LGT)

LET R2 = 642 / (LGHT.D2-FAILUME - FSAF.LGT)

LET R4 = 642 / (LGHT.D2-FAILUME - LGT)

LET R4 = 642 / (LGT)

LET R4 = 642 / (LGHT.D2-FAILUME - LGT)

LET 
                                                               LET v(2) = ICL * (2.0 * *FR(ID) + MSR(ID)) * CGRI(ID)
LET x(3) = T&Q - IIME,V
LET v(5) = v2 + (x(3)-x(2)) * (3.0 * *FR(ID)) * CGRI(ID)
LET v = 3
HFME
IF v GE 5
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        FISE
SCHFDULE A PEACH, FAIL, SAFE, LGT (ID) AT TIME, V + T2
LET AMFSL(IG) = PFACH, FAIL, SAFE, LGT
LET ALT (IG) = "YES"
PFGAPPILESS
LET AT A + FXP, F(SU + M)
LET AT A + RP RIVER (ID)
LET AT A + RP RIVER (ID)
LET TE = "LOG, E, F(KPN(ID))
LET TE = "LOG, E, F(KPN(ID))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      LEI GAZ = ($4xeckt - mtenk) / ($PhankS - kkaa2)
PEGAHDLESS
LEI 12 = II + (FSaF_LGI - (GH1a11))/GHZ
IF ICL UF CCL
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                LET - x = x + x(1) = x(1)

LET - x = xx + x(1) = x(1) = x(1)

LET - xx = xx + x(1) = x(1)

LET - xx = xx + x(1) = 
CACT SIMSCAIPT 11.5 RELEASE AF
                                                                                                                                                                                                                                                                                                                             LET 4(1) = 100.0
FIN I = 2 IN N
                                                                                                                                                                                                                                                                                                                                                                                                                                         LET A(1) = 1.0
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    FOR 1 = 1 Th 4
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LINE CACI STUSCHIPT II.S PFLEASE ME

	ACIR0105	PA CHICA	40 0101	E A PARTY	FURD RS	0	2	F(3K) 74	#CMD 73	17 00.00		F(34.5)	FILED A7	42	tering .	RINK! 31	P(141) 29			\$ C. P. C. P.	nukb 67	ACM CHIS	4 6 6	ACRO S	FC#0 50	MURD. 39	•	
	A 1811G	9 10110	ממממ	DAIUHLE	A Millor	1 00000	מנונים	INTEGER	7. E. K.	1 1111111	ואוווונו	OCCUBLE	4 101.474	17000	DOUBLE	Partition F	3 41.10	10000	DOUBLE	26.43	3 181110	1	The Court	DOUBLE	Pulling &	4 10100	10000	
SF AF	•	-	<b>1</b>	<b>7</b>		دو	LIST			¥	÷:	2		-8	4.			ננ		• ,	•	433	٠٠	*		2	ر کز	
II.5 RELEASF AF		#OR0103	A 05.30		C (17)	11 0204	10 000		#C4 07	Belief 18	24 (4.1)		A()R() 71	50 070		50 67 68	٦ (١٩٠)	11 0000	7	204 F.204	#040 St	10 0214	14 67	A CO C. MALLEO	#()#1) <b>3</b>	1001	#DR0 37	
CACI SIMSCRIPT		DOLLER P	3 10137.4	District Confession	OCCUPILE.	CALL BEALE		DOWNER	SOUPLE	4 agricus		ひばいが	PAURLE	1	משמנו	ocuett	ODER E		DOOMER	DOUBLE	COURLE	OCHINE E		שבוני	WEAL	06 At	A SERVICE	74.500
TAG CAGI		•	214	Ç	4		٠.	1.6KS			¥.	23		'n	25		•	¥ 4	182	1	: :	<b>y</b> :	4	¥		4 )	_ {	<u>ដ</u>

-/42

09/01/76 P4GE dc	Salve Inches Aho occase Edicat michaes	
Laci Stascarpr 11.5 AFLFASF MF  Event 1.11E(10)1E)  DFF1-4 Diff AS AG TAFEGER WANTAHLF  LF 1.17F(10) = "cq"  LF 1F1(10) = "cq"  LF 1F1(10) = "cq"	First I = 1 Fit WidC(Lizz)  As to 1 interpolation   1 interpolatio	Lucke wagraues of imis aucrius  [-15:4- 4:40] 7
	# · 2	-/43 -

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3 2 6						in Li				
867.17.70 PAGE 45						:				
198					Through a stream	の時 「日野の年出版は e ・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・				
LINE CACI SIMSCHIPT II.5 WELFASE AF	2.IFC:026) OFFILE 1924 as an Isteer variable Let In = 1926	[FT 7.1.1([5]) H #45.8 [FT [F7.0]] H #55.8 [FT-H #4F.84	$\frac{F(W)}{2} = \frac{1}{2} + \frac{1}{2} + \frac{W(S)}{2} = \frac{1}{2}$	In a full-flowers	PHINI I OHRHUE LINE AITH IN CHINAMALLINHECTA IINGAR FUTAY,IIME(AIAPLAME(IDI) AS F(ELLOS	AZC 2012, www.abb Interfebbb SECCOMP Cabe test is extended at meles indicated and				
CACI SIMSCHIPT	EVENT ZIJTE (1926) OFFINE JOPE OFFINE TO = 19	LFT 2-11-1(TD) LFT TF-2(TD) = TF-	F.14 1 = 3	SKIP ILITED TO SELECT	o I ivind	470 %). LF38E	ttst Trib	RIARYS	HE T-14P.	FND
LILE	<b>~~</b> ~	<b>a</b> v ¢	~ «	<u>.</u>	= 2	1.3	<del>1</del> 1	<u>.</u> <u>.</u>	: 1	×

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(14)4			11 1 705
1:16 SE +	19.78@FE	1.17.1.1.1	Isteben
	10 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	add 7 I.a little right r	a137) 7 1.2 1.1514 r.m.n. a137) 9 10.24 10.15145 r.m.n. a132) 10 4.1 10.15145 r.m.n. a133) 13 4.5 10.1516 r.m.n. a143) 15 1.3 10.1516

The continue of the continue o

09/01/76 FAGE 46	10 = 7LIO(LOX.1) 10 = 7LIO(LOX.1) 11 10104LF LINE 11 10 10 10 10 10 10 10 10 10 10 10 10 1	
<b>0</b>	6 v -	
	# 34 # # # .	<b>60 ~~ √</b> 13 <b>~</b> 1√
	T*FCCL.	KURD B KÜRD 12 FÜRD 12 FÜRD 14
Plakle	CM11.CM.LG GLLUMS IRD CMACK M	ITINE INTEGEN INTEGEN INTEGEN INTEGEN
3.ITE(103E) 3.ITE(103E) DEFINE 103E AS AN INTEGEW VARIABLE LET 10 = 103E LET 15.INT(10) = "NO;" LET 15.SILD) = "NO;" I HI] = "YES." FINE 1 = 1 17 NOBELLOS.	DU = 1LID(LDX,I)  SAIP I NUPOT LINE PHINT I NUPOT LINE ENIMY, IIME (AIRPLANFID)) AS FULLUAS A/C MU, N.S. MAS INTERMAL IMIRD CHACK BFCIME EXTERNAL A SF LOOP AVS WETHER	LUCAL VARTARLES OF THIS MOUTINE FOER AIM T 1.2 IN EGER WIND 10 K.1 IN EUGH WIND 13 K.1 IN EUGH WIND 15 K.1 IN EUGH WIND 10 IN K.1
3.TE(103E) DEFINE 103E AS AN INTEGENTE 103E AS AN INTEGENTE 103E LET 3.INT(10) = "NO" HALL TET [E.S. (10)] = "NO" HALL TENES = "NO" HALL TET [E.S. (10)] = "NO"	IF DD = LLD(LOX, 1) SKIP 1 NUTED LINE PHINT 1 DOUGLE LINE ENIMY, TIME (AIRPLANE A/C MI), A** HAS N LEAVE LLAVE LLAY LLAY LLAY RLS LLAY FUL	MAGINALES  MINAL  MINAL
EVENT 3.TE(103E)  DEFINE 103E  LET 10 = 103E  LET 3.NOTOD  LET 11.NOTOD  IF 11.NOTOD  F 11.NOTOD  IF 11.NOTOD	IF 10 = 1 SK 10 = 1 SK 10 = 1 SK 10 = 1 FK 10 + 1 LEAVE FLSF LLANE FLSF ALMAYS WETURN FWU	LUCAL Thirthe Integen Integen Integen
~~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~ ~	20 C C C C C C C C C C C C C C C C C C C	-/65

4.7 SCHENGLES INSPECTIONS ALTO OFFINAL LEVEL ON A GIVEN DIFICHAFT SUCH THAT THE AIMCHAFT IS WIT THEOLOGIEN PHICH TO CHACK AFACHING MINIMUM FFIFCIAMLE LGT 04/01/16 LFI S.1.39.at = FNRY\_IIVE(AIRDLANCID)) + C.INTERVAC(D))
- FUNC\_F(CTOL=FNINY\_IINF(AIRDLANF(ID))/C.INTERVAL(D) + 1.0)
- SCHFOULF = C.LF\*EL\_INSPECTION(ID) at S.INSP\_at
- RT afc(ID) = C.LF\*EL\_INSPECTION
- MF1AMPLESS offide a \$5 av [NTFoff variable]

LET C1 = C.GeNatH.Paif a CH4F

LET C1 = C.GeNatH.Paif a CH4F

LET LYSP.LEVEL LET

LET IVE = 114f\_v + .551/v)

LET IVE = 114f\_v + .994o/t|

LET S.145P\_AT = FYTHY\_TIVE(A14PLANE(ID)) + 1a+Co(t)

\* IMML\_F([IM\_FVIFY\_IIVE(A14PLANE(ID))) / 1a+Co(t)

\* IMML\_F([IM\_FVIFY\_IIVE(A14PLANE(ID))) / 1a+Co(t)

\* IMML\_F([IM\_FVIFY\_IIVE(CT)) / 1a+Co(t)

\* I LFT S.INSW.AT = EWINT.[INF(AIMPLANF(IR)) + IAHUB(2)

\* TRUNT.F(ITML-EMIGY.FINF(AIMPLANF(IR)))//IAHUB(2) + 1.0)
SCHFBULF A 4.LEVFL.IMSPF(ITM.(IR) AT S.INSP.AT
LFT AML(IW) = H.LEVFL.IMSPF(ITM) LFT TVL = 11°F.V + .545/() IF PXT\_PISH\_LEWFL\_LE 4
LET IML = IIMF,V + .264/M1
IF H = 2 LET 12 = 114E.V + .563/C1 MINITINE INSPECTIVIESCHEDULEM(C.) LET INSP.SCH(ID) = "YES" METHRA FHD CAUL SIMSCHIPT II.S MELEASE WE WAPPLESS JE GAHPLESS -166-

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CALI EXANT FILLOPPING STANNESSENCIÓN STELOTOS PIDED ANTIONA STAL AN SATERIARA DESIGN +vex1 D.LEVEL.INSACCITY. (10A)
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LET 10 = 10A
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(+3 \*SVL = 1.0) Cact Stassalpt 11.5 46162SE ME

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1 EVENT MALEVEL, TASPECTION (108)
2 LET 1D = 1D a
3 I LET 1D = 1D a
4 I TE ERT, TASPALEVEL LF 1
5 LET ALL VEL, TASPECTION = AAL(1D)
6 LET AAL(1D) = AALEVEL, TASPECTION
7 CANCEL THE AAL(1D) = AALEVEL, TASPECTION
8 ALPAYS
10 ALPAYS
11 LET AAL(1D) = AALEVEL, TASPECTION
11 LET AAL(1D) = AALEVEL, TASPECTION
12 LET ASYP = 1.0
13 CALL ERAMIAF(2, AAPAIA, CUST
14 CALL ERAMIAF(2, AAPAIA, CUST
15 SEMPOULE AALEVEL, TASPECTION(1D) AT TIME, V + 1 AACD(2)
16 CALL ERAMIAF(2, AAPAIA, CUST
17 LET AAL(1D) = AALEVEL, TASPECTION(1D)
18 SEMPOULE AALEVEL, TASPECTION(1D)
19 ET AAL(1D) = AALEVEL, TASPECTION
19 ET UNH
19 END

TRIFGER MUND

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	EVENT REACH, FAIL, SAFE, LGT (10HFS)	153						
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IF ID = ILID(LDX,I)

SKIP ( OUTP-I LIVE

PRINT I LIVE VITH ID, TIME,V-FMTHY,TIME (AIPPLANE (ID)) AS FULLONS

A/C MI, A+A EXPERINCES ELEMEN! FAILURE AT \*\*\*\*\* FLIGHT HOURS

LET I,STRENGTH,WEDUCTION = AISH(IN)

LET IA = IIVE,A(I,STMENLTH,WEDUCTION)

LET CCL = CRIT-CRN,LGI

IF Ccl,EXISTS(IN) = "YES"

LET COMMISSION = AC(IN)

LET COMMISSION = AC(IN)

LET COMMISSION = AC(IN) | FF | ICL = (TAC-TA1)AMSR(1D) + (TIMF,V=TAC)AMSR(1D)ACGRI(1D) ALMAYS | F | ICL G| CCL | FF | ICL = CCL + (ICL-FCL)A(WFW(1D)/HSR(ID)) ALMAYS IF 2.cd.Exists(ID) = "\*fs"
LET 2.STEE4GH.REDICTION = a2SR(ID)
LET 342 = 11 \*F. a(2.STEE4GT \*F. BEDUCTION)
LET 2CL = (TIME.v=Ta2)\*\*SR(ID)\*CGMT(ID)
IF 1aC GT 1a2
LET 2CL = (TaC=1a2)\*\*SR(ID) + (TIME.v=TaC)\*\*SR(ID)\*CGRI(ID)
al. ars
IF 2CL = (TaC=1a2)\*\*SR(ID) + (TIME.v=TaC)\*\*SR(ID)\*CGRI(ID)
al. ars
LET 2CL = CCL + (2CL-CCL)\*(MFH(ID)/USR(ID)) LET SCL = (Tac-tas) + vS+(10) + (11×E,v-tac) + vS+(10) + CG+(10) + (11×E,v-tac) + vS+(10) + CG+(10) + (11×E,v-tac) + vS+(10) + (11×E,v-tac) + LFI CL = 1CL+2CL+5CL LEI S = 2.75 - CL+(1.75/FSAF.LGI) IF CL GI FSAF.LGI LEI S = (L641.70.FAILUMF-CL)/(LGH1.10.FAILUME-FSAF.LGI) AL\*AYS PHIST 2 LIMFS AITH CL, 5/2.75 AS FULLGAS ALAAYS

1F 5.CA.FRISTS(10) c "YES"

LET 5.SIRFRGIM.REDUCTION = ASSM(10)

LET 1A3 x 11AE.A(5.SIREMIN.MEDUCTION)

LET 3CL x (TIMF.V\*TA3) aMSM(10) \*CGMI(10) LEI ICL = (TIMF.v-TAI).\*SH(IL).CGHI(ID) IF TAC GT TAI EVENT FAILURE (IDFA)

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IF ITHO = "PFS"

FOR I = 1 TO MORC(10X) IF TAC GT TAS ç 22522

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LFT STSEL = SWSEL + 1

LFT SWCIDGSWSEL) = 10

LFT SSTAMGSWSEL) = 119E.V + FLIPY.IIMF(AIGPLANE(10))

LFT SSTAMGSWSEL) = FLFWEQT(4)

FREWARDLESS

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CACI SIMSCRIPT II.S WELFASF AF

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MEGAROLESS	LET D.LEVEL. INSPECTION = ADL(ID)	CANCEL THE D.LEVEL. INSPECTION	DESTRUY THE DALEWEL INSPECTION	1F A1L(10) = "4FS"	LFT AFACH.FSIL.SAFF.LGI = AMFSL(TD)	CANCEL THE REACH FAIL SAFE LG!	DESTANY THE MEACH, FAIL, SAFE, LGT	LET AIL(IU) = "NO"	RFGAMPLESS	JF +5+(10) = "YES"	LET FAIL : AF(10)	CANCEL THE FAILURE	HESTHIY THE FAILURE	LET FSH([·] H "AG"	AL 4AYS	-	AP T1944	Chi
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CANCEL THE FAILURE

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1F 12C G1 TA1

LE1 CL = (13C-TA1)*MSR(ID) + (TIME.V-TAC)*MSR(ID)*CGHJ(TD)

KEGARDLESS

IF CL LT CL

LET CL = CCL + ((CL-CCL)/MSM(ID))*MFM(IU)

HEGARDLESS
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LET FOTC.C44 = VAX.C44 + 11.YEAN + A4C.(LIL)

1F FOT.C44 +1 CCL

LET FOT.C44 = CCL + (FF1CRA-CCL)/VI.MEA.) + V2.VEAN

HEGAROLESS
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CACI SIMSCHIPT 11.5 HELFASE NE

IF FLEET.STR.RED GI (FSAF.LGI75.0) \* (IDCN-1.NUM.DF.REJIRE-2.VUM.DF.CRASH)
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IFT, ACCEL. TATE OF THE FORM TO THE TO THE V. OF STANTIFEL + ACTUAL, AVG. FAILLIFF /

ALTOL. AMERICAL TATE OF THE STANTIFE = "NO"

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LFT FMIMOLIFF = SECOMBLIFF
LFT SECUMBLIFF = TYPELESSIMMETH. FOURTION - IIME, V. CALEE THE SSIME SECOMBLE SEC IF \$.f4.ex[SIS(I)) = "45"

IF 1146.a63.SIMEDUCIIUN) = TIME.V || FIMSI.LIFE

LET GATRO\_LIFE = SECOROLIFE

LET SECONOLIFE = FIRSI.LIFE

LET FERST.LIFE = FIRST.LIFE

LET FERST.LIFE = FIRST.LIFE

LET FERST.LIFE = FIRST.LIFE

LET SECOROLIFE = TIME.A65.SIMEDUCIION

DESTAND IME \$.SIMENGIM.MEDUCIION LET IMIMOLLIFF = IIME.A(\$,\$TREPSIM.REDUCTION) + IIME.v CANCEL IME \$,\$IMFOGIM.MEDUCTION L+1 AAFL = ACTUAL, AvG, FAI, LIFF TF I . LE FOCK Ť CACI SI +SCHI + 11.5 HELFASE LET ABEL = TABEL MEGAPOLESS Just Anta. JULY AMEAN アグルー・アカウルギ - 190 -

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IF 2.CP.EXISIS(ID) = "NS"

IF 5.CP.EXISIS(ID) = "NS"

IF 114L.A(5.5TREPGTH.XFDOCTION).- IIVE.V LT FIRST.LIFE

LFT THIRD.LIFE = FIRST.LIFE

LFT SECOND.LIFE = FIRST.LIFE

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LET SECHOLLIFF = FIRST.LIFF

LET SECHOLLIFF = TIRST.LIFF

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PESTWIY THE 2.STRENGTH.4EDUCTION
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LET INIKO, LIFE = TIME, ALZ, STHENGTH, HEDUCTION)
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LET 2.CCA.FRISTS(1D) = "NN"
LET 5.CCA.FRISTS(1D) = "NN"
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IF AACOILLIDAS,FOFG,CMS GF IABCDILLAID

LET OTFF = AACOILT) - AMCDILLDAS,FMFQ,CMG

LFT AACOILT) = AACOILTLDAS,FMFQ,CMG
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LET C.LEVEL.INSPECTION = ACL(IC)
CAMCEL THE C.fFVEL.INSPECTION
IF TINE.ACC.LFVEL.INSPECTION - TIME.V LT DIFF
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LFI AMCD(LEL) = AMCD(LFL) + S.FREG.CMG
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          EVENT INCREASE, INSPECTION, FREDUENCY
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CACT STUSCAPPT IT.S RELEASE AF
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CACI SIMSTHIPT II.S WELFASE AF

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IF WAR, CR ST CCL

IF WAR, CRS = MAX, CRS + 42, MAN + ASCR(LL)

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IF DIT, ST4, 4En = DIT, C4K + ST4, 42) + MAX, CRS

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LET APPL = TRING, FILEAD, TIME/PRODUCTION, TIME)

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